

AD-A254 810



WL-TM-92-338-FIGC

①

X-29 LINEAR AERODYNAMIC PERTURBATION MODEL

THOMAS A. GENTRY
Flying Qualities Group
Control Dynamics Branch

DTIC
ELECTE
SEP 3 1992
S c D



July 1992

Approved for Public Release; Distribution Unlimited

FLIGHT DYNAMICS DIRECTORATE
WRIGHT LABORATORY
AIR FORCE MATERIEL COMMAND
WRIGHT-PATTERSON AFB, OHIO 45433-6553

42-24301
152
422730

92 9 02 210

Foreword

This work was performed by Thomas A. Gentry of the Control Dynamics Branch, Flight Control Division, under job order number 24030595.



FRANK R. SWORTZEL, CHIEF
Control Dynamics Branch
Flight Control Division

DTIC QUALITY INSPECTED 3

Accession For	
NTIS GRA&I	<input checked="checked" type="checkbox"/>
DTIC TAB	<input type="checkbox"/>
Unannounced	<input type="checkbox"/>
Justification	
By	
Distribution/	
Availability Codes	
Dist	Avail and/or Special
A-1	

INTRODUCTION

This memo documents the linear aerodynamic and thrust perturbation model of the X-29 simulation model resident in the Control Integration and Assessment Branch of Wright Laboratory (WL/FIGD) at Wright-Patterson Air Force Base, Ohio. This model was used to perform flying qualities evaluations of the X-29 aircraft at Wright Laboratory during the airplane's development in the early to mid-1980s.

This work was conducted in order to document the X-29 longitudinal and lateral-directional linear aero model at numerous trim flight conditions for use in future research work in flying qualities. The X-29 is representative of typical fighter aircraft, even with its forward-swept wing, although the airframe is very unstable longitudinally.

This memo is divided into four sections: Introduction, Definition of Terms, Explanation of Computer Printouts, and Computer Printouts sections. Definitions of all terms associated with the linear model are given in the Definition of Terms section. Explanation of all of the data in the computer printouts is given in the Explanation of Computer Printouts section. The Computer Printouts section contains the actual computer printouts for each trim flight condition.

Mr Dan Goddard of WL/FIGD wrote much of the original computer software that generated the linear model given in this memo. The software was modified where necessary to make the computer printout easier to understand and to complete the lateral-directional section of the software. Mr Brian Stadler and Mr Joe Nalepka of WL/FIGD resurrected and documented the X-29 aero model for this project and helped immensely in explaining the computer system and software at WL/FIGD.

DEFINITION OF TERMS

The aerodynamic forces and moments for the model are defined in terms of nondimensional aero coefficients, C_x , C_y , C_z , and C_l , C_m , C_n , respectively. The x axis is positive out the front of the airplane, the y axis is positive out the right wing, and the z axis is positive out the bottom of the airplane. The aerodynamic forces and moments are functions of twelve variables, seven in the longitudinal axis and five in the lateral-directional axis. These twelve variables are: angle of attack, pitch rate, Mach number, angle of attack rate, strake deflection, canard deflection, flaperon deflection, angle of sideslip, roll rate, yaw rate, aileron deflection, and rudder deflection. Thrust variation is also included in the linear model and is a function of two variables, Mach number and altitude. Thrust is aligned along the aircraft centerline parallel to the x axis (no inclination angle) above the center of gravity (negative pitching moment).

The linear model is produced by perturbing each of the twelve variables, one at a time, in a positive direction and then in an equal negative direction from the trim value. The partial derivatives required for the nondimensional and dimensional derivatives are calculated by dividing the total variation in each nondimensional aero coefficient or thrust by the total variation in each of the twelve independent variables. Altitude is not

varied in these linear models. All angular and angular rate independent variables are varied plus and minus one degree or one degree/second, respectively. Mach number is varied plus and minus 0.05.

All aero coefficients and nondimensional and dimensional derivatives follow the definitions given in Appendix A of NASA CR-2144 (Reference 1) except for two variations. It was decided to make all sign conventions agree for forces and moments. This is so all nondimensional coefficients and nondimensional and dimensional derivatives are positive in the positive directions of the axis system. Therefore, C_x is the opposite sign of the C_x in NASA CR-2144 and C_z is the negative of the C_N in NASA CR-2144. The lateral-directional L and N dimensional derivatives are converted to primed derivatives as defined in NASA CR-2144 for print out. Please note that all coefficients and derivatives are in a body axis system, not a stability axis system.

The dimensional derivatives are used to generate a longitudinal and lateral-directional state space model according to the following equations. The general form of the state space model is:

$$\dot{x} = Ax + Bu$$

where A is a 4x4 matrix and B is a 4x3 matrix for the longitudinal model or a 4x2 matrix for the lateral-directional model.

Longitudinal State Space Model

The state vector x contains forward speed, angle of attack, pitch rate and pitch angle. The input vector u contains strake, flaperon and canard deflections. The elements of the A matrix are:

$$A(1,1) = X_u^* + \frac{X_{\dot{\alpha}} Z_u^*}{U_0 - Z_{\dot{\alpha}}}$$

$$A(1,2) = X_{\alpha} + \frac{X_{\dot{\alpha}} Z_{\alpha}}{U_0 - Z_{\dot{\alpha}}}$$

$$A(1,3) = X_q - W_0 + \frac{X_{\dot{\alpha}}(U_0 + Z_q)}{U_0 - Z_{\dot{\alpha}}}$$

$$A(1,4) = -g \cos \theta_0 + \frac{X_{\dot{\alpha}}(-g \sin \theta_0)}{U_0 - Z_{\dot{\alpha}}}$$

$$A(2,1) = \frac{Z_u^*}{U_0 - Z_{\dot{\alpha}}}$$

$$A(2,2) = \frac{Z_{\alpha}}{U_0 - Z_{\dot{\alpha}}}$$

$$A(2,3) = \frac{U_0 + Z_q}{U_0 - Z_{\dot{\alpha}}}$$

$$A(2,4) = \frac{-g \sin \theta_0}{U_0 - Z_{\dot{\alpha}}}$$

$$A(3,1) = M_u^* + \frac{M_{\dot{\alpha}} Z_u^*}{U_0 - Z_{\dot{\alpha}}}$$

$$A(3,2) = M_{\alpha} + \frac{M_{\dot{\alpha}} Z_{\alpha}}{U_0 - Z_{\dot{\alpha}}}$$

$$A(3,3) = M_q + \frac{M_{\dot{\alpha}} (U_0 + Z_q)}{U_0 - Z_{\dot{\alpha}}}$$

$$A(3,4) = \frac{-M_{\dot{\alpha}} g \sin \theta_0}{U_0 - Z_{\dot{\alpha}}}$$

$$A(4,1) = 0$$

$$A(4,2) = 0$$

$$A(4,3) = 1$$

$$A(4,4) = 0$$

The elements of the B matrix are:

$$B(1,1) = X_{\delta_s} + \frac{X_{\dot{\alpha}} Z_{\delta_s}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(2,1) = \frac{Z_{\delta_s}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(3,1) = M_{\delta_s} + \frac{M_{\dot{\alpha}} Z_{\delta_s}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(4,1) = 0$$

$$B(1,2) = X_{\delta_r} + \frac{X_{\dot{\alpha}} Z_{\delta_r}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(2,2) = \frac{Z_{\delta_r}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(3,2) = M_{\delta_r} + \frac{M_{\dot{\alpha}} Z_{\delta_r}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(4,2) = 0$$

$$B(1,3) = X_{\delta_o} + \frac{X_{\dot{\alpha}} Z_{\delta_c}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(2,3) = \frac{Z_{\delta_o}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(3,3) = M_{\delta_o} + \frac{M_{\dot{\alpha}} Z_{\delta_o}}{U_0 - Z_{\dot{\alpha}}}$$

$$B(4,3) = 0$$

Terms with subscripts of δ_s , δ_f , or δ_c are associated with the strake, flaperon, or canard control surfaces, respectively.

Lateral-Directional State Space Model

The state vector x contains angle of sideslip, roll rate, yaw rate, and bank angle. The input vector u contains aileron and rudder deflections. The elements of the A matrix are:

$$A(1,1) = \frac{Y_{\beta}}{V_{T_o}}$$

$$A(1,2) = \frac{Y_p + W_0}{V_{T_o}}$$

$$A(1,3) = \frac{Y_r - U_0}{V_{T_o}}$$

$$A(1,4) = \frac{g \cos \theta_0}{V_{T_o}}$$

$$A(2,1) = L'_{\beta}$$

$$A(2,2) = L'_p$$

$$A(2,3) = L'_r$$

$$A(2,4) = 0$$

$$A(3,1) = N'_{\beta}$$

$$A(3,2) = N'_p$$

$$A(3,3)=N_r'$$

$$A(3,4)=0$$

$$A(4,1)=0$$

$$A(4,2)=1$$

$$A(4,3)=\frac{\sin \theta_0}{\cos \theta_0}$$

$$A(4,4)=0$$

The elements of the B matrix are:

$$B(1,1)=\frac{Y_{\delta_a}}{V_{T_0}}$$

$$B(1,2)=\frac{Y_{\delta_r}}{V_{T_0}}$$

$$B(2,1)=L_{\delta_a}'$$

$$B(2,2)=L_{\delta_r}'$$

$$B(3,1)=N_{\delta_a}'$$

$$B(3,2)=N_{\delta_r}'$$

$$B(4,1)=0$$

$$B(4,2)=0$$

Terms with subscripts of δ_a or δ_r are associated with the aileron or rudder control surfaces, respectively.

EXPLANATION OF COMPUTER PRINTOUTS

The computer printout is divided into six pages for each trim condition. The first page is a printout of the trim condition including the aircraft state, positions of the controls, the trim aerodynamics, and aircraft constants. The second page is a printout of all of the nondimensional aero derivatives and thrust-related derivatives. The third page is a printout of the longitudinal dimensional stability derivatives, the longitudinal state space

model, the roots of the longitudinal characteristic dynamic equation, and the period, damping ratio, damped and undamped natural frequencies for each complex pair of roots present in the characteristic equation. The fourth page is a printout of all longitudinal transfer functions for each state output to each control surface input and the transfer function for normal acceleration output at the center of gravity to each control surface input. The fifth page is a printout of the lateral-directional dimensional derivatives, the lateral-directional state space model, the roots of the lateral-directional characteristic dynamic equation, and the period, damping ratio, damped and undamped natural frequencies for each complex pair of roots present in the characteristic equation. The sixth page is a printout of all lateral-directional transfer functions for each state output to each control surface input and the transfer function for side acceleration output at the center of gravity to each control surface input.

The first page summary of the trim condition is fairly self-explanatory. All trim conditions are for straight, wings-level, 1g flight. All angular values are in degrees. The variable "VTRFPS" is the aircraft trim airspeed in feet per second. The fuselage station and waterline position of the center of gravity are given as the variables "XCG" and "ZCG", respectively, and are in inches. The flight control system MODE status logicals at the bottom of the AIRCRAFT STATE section are a carryover from the original linearization software code and have no effect on the linear model given in the printout. The model is strictly for the airframe aerodynamics and thrust variation with Mach. Based on the signs of the nondimensional and dimensional derivatives, the three longitudinal controls, canards, flaps, and strakes are positive trailing edge down. The rudder is positive trailing edge left and the ailerons are positive right wing surface trailing edge up. The thrust at trim is shown in pounds. Also, although not printed out, the thrust line is above the center of gravity by a distance of 75.204 inches minus the value of "ZCG" (negative pitching moment). The variable "% RPM" is the percent of full throttle required to set the thrust at the trim value. The nondimensional aero coefficients at trim are given in the DYNAMICS section and are positive in the positive directions of the axis system as explained earlier.

The second page of the printout is a listing of all of the nondimensional aero derivatives and thrust variation with Mach perturbation. The calculation of the nondimensional derivatives and thrust variation is consistent with Appendix A of NASA CR-2144.

The third page of the printout lists the longitudinal dimensional stability derivatives, with units shown. The thrust variation with Mach perturbation has been included in the calculation of these dimensional derivatives. The longitudinal state space model A and B matrices are then printed out. All angular states, "ALPHA", "Q", and "THETA", and the control surface inputs, "STRAKE", "FLAPRN", and "CANARD" in this model have units of radians or radians/second, as appropriate, and the speed state, "VEL", has units of feet/second. The system roots of the A matrix are then printed out in real/imaginary form. Finally, the period, damping ratio, and damped and undamped natural frequencies for any complex pair of system roots is printed out.

The fourth page of the printout lists the transfer functions in Laplace form for the

output of each state to each control surface input. The units on the transfer functions are the same as the units on the states and surface deflections in the state space model. Also, the transfer function of the acceleration at the center of gravity in the z direction to each control surface input is also listed as "AN/STRK", "AN/FLAP", AND "AN/CAN". These transfer functions are based on the states according to the following equation:

$$AN = U_0 \dot{\alpha} - U_0 q + (g \sin \theta_0) \theta$$

The units on this acceleration are feet/second² and it is positive in the positive z direction.

The fifth and sixth pages repeat the same type of information for the lateral-directional motion as the third and fourth pages did for the longitudinal motion. The lateral-directional dimensional derivatives have units as shown. The lateral-directional state space model states and surface deflections all have units of radians or radians/second, as appropriate. The transfer functions include the acceleration at the center of gravity in the y direction for aileron and rudder deflections, "AY/AIL" and "AY/RUD", respectively. The transfer functions are based on the states according to the following equation:

$$AY = U_0 \dot{\beta} - (g \cos \theta_0) \phi + U_0 r - W_0 p$$

The units on this acceleration are feet/second² and it is positive in the positive y direction.

Computer Printout Variables By Page

Page 1

The variables printed out on page 1 are as follows:

ACGW	Aircraft gross weight, pounds
AREA	Reference wing area, feet ²
AILERON	Trim aileron deflection, positive right wing surface trailing edge up, degrees
ALPHA	Trim angle of attack, positive nose up, degrees
ALT	Altitude above sea level, positive up, feet

AR MODE	Flight control system mode, not used
BETA	Trim angle of sideslip, positive nose right, degrees
CANARDS	Trim canard deflection, positive trailing edge down, degrees
CBAR	Reference chord, feet
CLMAERO	Trim nondimensional x-axis aero moment coefficient, positive right wing down
CMMAERO	Trim nondimensional y-axis aero moment coefficient, positive nose up
CNMAERO	Trim nondimensional z-axis aero moment coefficient, positive nose right
CXAERO	Trim nondimensional x-axis aero force coefficient, positive forward
CYAERO	Trim nondimensional y-axis aero force coefficient, positive out the right wing
CZAERO	Trim nondimensional z-axis aero force coefficient, positive out the bottom of the airplane
DR MODE	Flight control system mode, not used
DYN PR	Trim dynamic pressure, pounds/feet ²
FLAPS	Trim flaperon deflection, positive trailing edge down, degrees
GAMMA	Trim flight path angle, positive up, degrees
GEAR DOWN	Landing gear up if F(false), down if T(true), always F
IXX	Aircraft x-axis moment of inertia, slug-feet ²
IXZ	Aircraft product of inertia, slug-feet ²
IYY	Aircraft y-axis moment of inertia, slug-feet ²
IZZ	Aircraft z-axis moment of inertia, slug-feet ²
MACH	Mach number, dimensionless
NORMAL MODE	Flight control system mode, not used

NX	Trim forward acceleration in g's, positive forward, dimensionless, always zero
NY	Trim side acceleration in g's, positive to the right, dimensionless, always zero
NZ	Trim normal acceleration in g's, positive out top of airplane, dimensionless
% RPM	Trim throttle setting, percent of full setting
RHO	Air density, slugs/feet ³
RUDDER	Trim rudder deflection, positive trailing edge left, degrees
SPAN	Reference wing span, feet
STRAKES	Trim strake deflection, positive trailing edge down, degrees
THETA	Trim pitch attitude, positive nose up, degrees
THRUST	Trim engine thrust, positive forward, pounds
VC KTS	Trim calibrated airspeed, knots
VTRFPS	True airspeed, positive forward, feet/second
XCG	Fuselage station position of the center of gravity, positive forward, inches
ZCG	Waterline position of the center of gravity, positive up, inches

Page 2

This page is fairly self-explanatory. All nondimensional stability derivatives are per radian for angular perturbations and per Mach number for the Mach derivatives. Thrust is separated from the aerodynamic derivatives as shown in the far right column. Thrust only varies with Mach number. Thrust variation with all other longitudinal perturbations is zero as shown. Thrust variation with Mach number has units of pounds per Mach number as shown.

Page 3

All dimensional stability derivatives have units as shown at the top of each column and at the left of each row. For example, the units for Z_{α} are feet/second² per radian. The "feet/second²" is shown at the top of the "Z" column just under the "Z" and the "per radian" is shown at the left next to the "ALPHA".

The state space model variables are as follows:

ALPHA	Angle of attack, positive nose up, radians
CANARD	Canard deflection, positive trailing edge down, radians
FLAPRN	Flap deflection, positive trailing edge down, radians
Q	Pitch rate, positive nose up, radians/second
STRAKE	Strake deflection, positive trailing edge down, radians
THETA	Pitch attitude, positive nose up, radians
VEL	X-axis velocity, positive forward, feet/second

The SYSTEM POLES are the roots of the characteristic equation of the A matrix in real, imaginary form.

The DYNAMICS variables for each pair of complex SYSTEM POLES are as follows:

PERIOD	Period for one cycle, seconds
WD	Damped natural frequency, radians/second
WN	Undamped natural frequency, radians/second
ZETA	Damping ratio, dimensionless

Page 4

This page lists all of the longitudinal transfer functions in polynomial form for each state output to each control surface deflection input. It also gives the transfer function for the z-axis acceleration at the center of gravity to each control surface input. The output and input variables are as follows:

ALP	Angle of attack, positive nose up, radians
AN	Z-axis acceleration at center of gravity, positive out bottom of airplane, feet/second ²
CAN	Canard deflection, positive trailing edge down, radians
FLAP	Flap deflection, positive trailing edge down, radians
Q	Pitch rate, positive nose up, radians/second

STRK	Strake deflection, positive trailing edge down, radians
THT	Pitch attitude, positive nose up, radians
U	X-axis velocity, positive forward, feet/second

Page 5

All dimensional stability derivatives have units as shown at the top of each column and at the left of each row the same as page 3.

The state space model variables are as follows:

AIL	Aileron deflection, positive right wing surface trailing edge up, radians
BETA	Angle of sideslip, positive nose right, radians
P	Roll rate, positive right wing down, radians/second
PHI	Bank angle, positive right wing down, radians
R	Yaw rate, positive nose right, radians/second
RUD	Rudder deflection, positive trailing edge left, radians

The SYSTEM POLES and DYNAMICS are listed the same as for page 3.

Page 6

This page lists all of the lateral-directional transfer functions in polynomial form for each state output to each control surface deflection input. It also gives the transfer function for the y-axis acceleration at the center of gravity to each control surface input. The variable names and units are the same as the state space model names and units on page 5. AY is the y-axis acceleration at the center of gravity. It is positive out the right side of the airplane and has units of feet/second².

References

1. Heffley, R. K. and W. F. Jewell, "Aircraft Handling Qualities Data", NASA CR-2144, December 1972.

COMPUTER PRINTOUTS

This section consists of the following Mach, altitude combinations:

Mach	Altitude (feet)	Page
.3	100	13
.5	100	19
.7	100	25
.9	100	31
.95	100	37
.5	10000	43
.7	10000	49
.9	10000	55
1.1	10000	61
.5	20000	67
.7	20000	73
.9	20000	79
1.1	20000	85
1.2	20000	91
.5	30000	97
.7	30000	103
.9	30000	109
1.1	30000	115
1.3	30000	121
.7	40000	127
.9	40000	133
1.1	40000	139
1.3	40000	145

AIRCRAFT STATE

MACH =	.30000	ALT =	100.00	VTRFPS =	334.97
NZ =	.99503	ALPHA =	6.42495	BETA =	.00000
GAMMA =	.00000	THETA =	6.42498	ACGW =	15926.00
IXX =	4568.00	IYY =	49429.00	IZZ =	52531.00
IXZ =	1827.00	XCG =	450.56006	ZCG =	64.89000
GEAR DOWN:	F	NY =	.00000	NX =	.00000
NORMAL MODE:	T	DR MODE:	F	AR MODE:	F

CONTROLS

CANARDS=	-3.87056	FLAPS =	9.54432	STRAKES=	-7.74468
AILERON=	.00000	RUDDER =	.00000		
THRUST =	1797.586	X RPM =	9.79640		

DYNAMICS

CXAERO =	-.00062841	CYAERO =	.00000000	CZAERO =	-.64324743
CLMAERO=	.00000000	CMMAERO=	.00871561	CNMAERO=	.00000000
VC KTS =	198.088	DYN PR =	132.95784	RHO =	.00236997

A/C CONSTANTS

AREA =	185.0490	CBAR =	7.2200	SPAN =	27.2000
--------	----------	--------	--------	--------	---------

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.5321831	-4.7935514	1.7297268	.0000000
PITCH RATE(PER RAD)	.0000000	-6.5870295	-7.2356968	.0000000
STRAKE (PER RAD)	.0354164	-.2180922	-.3314270	.0000000
MACH (PER M #)	.0192620	-.0059533	.0156562	31.5185547
ALPHA DOT (PER RAD)	.0000000	-.4724714	-.9685700	.0000000
CANARD (PER RAD)	-.0462081	-.2957808	.7754377	.0000000
FLAPERON (PER RAD)	.0360977	-1.4238043	-.5511451	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0915518	.2806405	-.0387872
ROLL RATE (FER RAD)	.0000000	-.0325567	-.3489300
YAW RATE (PER RAD)	.0000000	-.2125371	.1889825
AILERON (PER RAD)	-.2156202	.0809913	-.2325273
RUDDER (PER RAD)	.2488604	-.1437383	.0413010

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-242.3988647	6.1472998	26.1457520
ALPHADOT (PER R/S)	-.2499233	-.0370441	.0000000
Q (PER R/S)	-3.5285263	-.2802473	.0000000
VEL (PER FT/S)	-.1103693	-.0018412	-.0081129
STRAKE (PER RAD)	-10.8402023	-1.1910839	1.7603598
FLAPRN (PER RAD)	-70.7697144	-1.9807072	1.7942238
CANARD (PER RAD)	-14.7016869	2.7867708	-2.2967596

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.8113E-02	.2615E+02	-.3748E+02	-.3197E+02	.1760E+01	.1794E+01	-.2297E+01
-.3313E-03	-.7277E+00	.9887E+00	-.1081E-01	-.3254E-01	-.2125E+00	-.4413E-01
-.1829E-02	.6174E+01	-.3169E+00	.4004E-03	-.1190E+01	-.1973E+01	-.2798E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

B Matrix:

STRAKE	FLAPRN	CANARD
.1760E+01	.1794E+01	-.2297E+01
-.3254E-01	-.2125E+00	-.4413E-01
-.1190E+01	-.1973E+01	-.2798E+01
.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

1.9813757E+00	,J(0.0000000E+00)
-3.0114660E+00	,J(0.0000000E+00)
-1.1294723E-02	,J(1.3401318E-01)
-1.1294723E-02	,J(-1.3401318E-01)

DYNAMICS

PERIOD= 46.8848	WD= .13401
ZETA= .08398	WN= .13449

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
1.0526638E+00 S3
-5.9254580E+00 S2
-1.1623371E-01 S1
-1.0793537E-01 S0

NUMERATOR U/STRK
1.7603598E+00 S3
4.558196E+01 S2
3.6661224E+01 S1
3.4560318E+01 S0

NUMERATOR ALP/STRK
-3.2542206E-02 S3
-1.1875410E+00 S2
-1.2668468E-02 S1
-1.0561358E-02 S0

NUMERATOR Q/STRK
-1.1998785E+00 S3
-1.0796480E+00 S2
-2.3349337E-02 S1
2.4281780E-07 S0

NUMERATOR THT/STRK
0.000000E+00 S3
-1.1898785E+00 S2
-1.0796480E+00 S1
-2.3349337E-02 S0

NUMERATOR U/FLAP
1.7942238E+00 S3
7.0267410E+01 S2
1.0275743E+02 S1
8.8515823E+01 S0

NUMERATOR ALP/FLAP
-2.1245015E-01 S3
-2.0200968E+00 S2
-8.3314553E-03 S1
-8.2647763E-03 S0

NUMERATOR Q/FLAP
-1.9728374E+00 S3
-2.7666025E+00 S2
-3.5276704E-02 S1
1.1211432E-06 S0

NUMERATOR THT/FLAP
0.000000E+00 S3
-1.9728374E+00 S2
-2.7666025E+00 S1
-3.5276704E-02 S0

NUMERATOR U/CAN
-2.2967595E+00 S3
-1.0807091E+02 S2
-6.9788635E+01 S1
-5.7100922E+01 S0

NUMERATOR ALP/CAN
-4.4134356E-02 S3
2.7431936E+00 S2
3.4181856E-02 S1
3.1826764E-02 S0

NUMERATOR Q/CAN
2.7884054E+00 S3
1.7833910E+00 S2
4.8269980E-02 S1
-1.9394921E-08 S0

NUMERATOR THT/CAN
0.000000E+00 S3
2.7884054E+00 S2
1.7833910E+00 S1
4.8269980E-02 S0

NUMERATOR AN/STRK
-1.0832066E+01 S4
7.7807617E-01 S3
3.5087305E+02 S2
3.6957264E-01 S1
-8.4145546E-02 S0

NUMERATOR AN/FLAP
-7.0716599E+01 S4
-1.5730957E+01 S3
9.1102100E+02 S2
-9.6936321E-01 S1
-1.2738001E-01 S0

NUMERATOR AN/CAN
-1.4690653E+01 S4
-1.5049316E+01 S3
-5.7220630E+02 S2
9.4742680E-01 S1
1.7379308E-01 S0

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-54.2552490	-4.3316641	3.4245672
P (PER R/S)	.0000000	-2.1210060	-.0906070
R (PER R/S)	.0000000	1.1002455	-.0716664
AILERON (PER RAD)	-10.7173338	35.1205597	2.2532625
RUDDER (PER RAD)	12.3695278	5.4173527	-1.6427412

B Matrix:

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.1620E+00	-.1119E+00	-.9937E+00	.9545E-01	-.3200E-01	.3693E-01
-.4332E+01	-.2121E+01	.1100E+01	.0000E+00	.3512E+02	.5417E+01
.3425E+01	-.9061E-01	-.7167E-01	.0000E+00	.2253E+01	-.1643E+01
.0000E+00	.1000E+01	.1126E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.8475848E-01	+J(1.9579487E+00)
-1.8475848E-01	+J(-1.9579487E+00)
-2.0374527E+00	+J(0.0000000E+00)
5.2320469E-02	+J(0.0000000E+00)

DYNAMICS

PERIOD= 3.2091	WD= 1.95795
ZETA= .09395	WN= 1.96665

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
2.354644E+00 S3
4.494622E+00 S2
7.638439E+00 S1
-4.122957E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-3.199529E-02 S3	3.512056E+01 S3	2.253262E+00 S3	0.000000E+00 S3
1.620761E+00 S2	1.082325E+01 S2	1.852409E+00 S2	3.537428E+01 S2
2.340450E+00 S1	1.299149E+02 S1	1.456455E+01 S1	1.103185E+01 S1
4.940338E-01 S0	-1.397612E+00 S0	1.241144E+01 S0	1.315550E+02 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
3.692771E-02 S3	5.417352E+00 S3	-1.642741E+00 S3	0.000000E+00 S3
2.319601E+00 S2	-7.016735E-01 S2	-4.114731E+00 S2	5.232365E+00 S2
4.300054E+00 S1	1.126225E+01 S1	9.185932E-01 S1	-1.165026E+00 S1
-1.781851E-01 S0	-1.229155E-01 S0	1.091574E+00 S0	1.136570E+01 S0

NUMERATOR /

NUMERATOR AY/ AIL	NUMERATOR AY/ RUD
-1.065002E+01 S4	1.229183E+01 S4
-2.690918E+01 S3	2.224221E+01 S3
-1.410280E+02 S2	-7.930119E+01 S2
-2.098828E+02 S1	-1.384416E+02 S1
-2.238276E+01 S0	4.567697E+00 S0

AIRCRAFT STATE

MACH = .50000 ALT = 100.00 VTRFPS = 558.28
 NZ = 1.00047 ALPHA = 3.76924 BETA = .00000
 GAMMA = .00000 THETA = 3.76926 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IYZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.94386 FLAPS = .86039 STRAKES= -4.57629
 AILERON= .00000 RUDDER = .00000
 THRUST = 2451.833 % RPM = 13.24040

DYNAMICS

CXAERO = -.02055624 CYAERO = .000000000 CZAERO = -.23251498
 CLMAERO= .000000000 CMAERO= .00426820 CNMAERO= .000000000
 VC KTS = 330.176 DYN PR = 369.32739 RHO = .00236997

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	-.3105441	-4.7509108	1.5126143	.0000000
PITCH RATE (PER RAD)	.0000000	-6.7812767	-7.4659365	.0000000
STRAKE (PER RAD)	-.0317523	-.2093256	-.3201184	.0000000
MACH (PER M #)	.0072588	.0219631	.0359202	180.3588409
ALPHA DOT (PER RAD)	.0000000	-.4774358	-1.0246849	.0000000
CANARD (PER RAD)	-.0009524	-.2538229	.8558812	.0000000
FLAPERON (PER RAD)	.0381018	-1.4610481	-.5628066	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0334225	.265612	-.0526842
ROLL RATE (PER RAD)	.0000000	-.0082143	-.3753325
YAW RATE (PER RAD)	.0000000	-.1988654	.1484445
AILERON (PER RAD)	-.2162113	.0853983	.2216526
RUDDER (PER RAD)	.2257051	-.1312984	.0375782

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-657.2287598	15.0522022	42.3516083
ALPHADOT (PER R/S)	-4244111	-0.0659596	.0000000
Q (PER R/S)	-6.0543032	-4819348	.0000000
VEL (PER FT/S)	-0.0348100	-0.013081	-0.0139725
STRAKE (PER RAD)	-28.9012756	-3.1956730	4.3839922
FLAPRN (PER RAD)	-201.7247925	-5.6183786	5.2606630
CANARD (PER RAD)	-35.0449371	8.5440788	-0.1314949

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-0.1397E-01	.4235E+02	-0.3670E+02	-0.3210E+02	-4384E+01	-5261E+01	-0.1315E+00
-0.6244E-04	-0.1179E+01	.9884E+00	-0.3794E-02	-0.5184E-01	-0.3618E+00	-0.6286E-01
-0.1304E-02	.1513E+02	-0.5470E+00	-0.2499E-03	-0.3192E+01	-0.5595E+01	-0.8548E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-4.7486935E+00	+J(-0.0000000E+00)
3.0247660E+00	+J(-0.0000000E+00)
-7.9873800E-03	+J(-7.3838472E-02)
-7.9873800E-03	+J(-7.3838472E-02)

DYNAMICS

PERIOD= 85.0936	WD= -07384
ZETA= .10755	WN= .07427

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
1.7399006E+00 S3
-1.4330477E+01 S2
-2.1974552E-01 S1
-7.9095244E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
4.3839922E+00	S3	-5.1841483E-02	S3	-3.1922588E+00	S3	0.000000E+00	S3
1.2252643E+02	S2	-3.1845198E+00	S2	-4.5980253E+00	S2	-3.1922588E+00	S2
7.1826355E+01	S1	-4.2989496E-02	S1	-7.9998374E-02	S1	-4.5980253E+00	S1
1.4676512E+02	S0	-4.0370971E-03	S0	1.2087951E-05	S0	-7.9998374E-02	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
5.2606630E+00	S3	-3.6184257E-01	S3	-5.5945482E+00	S3	0.000000E+00	S3
1.9907420E+02	S2	-5.7328577E+00	S2	-1.2155059E+01	S2	-5.5945482E+00	S2
3.0473364E+02	S1	-6.1170671E-02	S1	-1.7650610E-01	S1	-1.2155059E+01	S1
3.8870386E+02	S0	4.2593367E-03	S0	2.1828833E-05	S0	-1.7650610E-01	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-1.3149488E-01	S3	-6.2861621E-02	S3	8.5482187E+00	S3	0.000000E+00	S3
-3.1660815E+02	S2	8.4136229E+00	S2	9.2460108E+00	S2	8.5482187E+00	S2
-2.5112436E+02	S1	1.0792059E-01	S1	1.5391314E-01	S1	9.2460108E+00	S1
-2.9437793E+02	S0	1.9312758E-02	S0	-3.1861404E-05	S0	1.5391314E-01	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-2.8879272E+01	S4	-2.0157123E+02	S4	-3.5018250E+01	S4		
4.3110352E+00	S3	-7.7047852E+01	S3	-7.4976563E+01	S3		
2.5307170E+03	S2	6.7252930E+03	S2	-5.0724648E+03	S2		
3.2590607E+01	S1	7.4990250E+01	S1	-5.5425919E+01	S1		
-1.7593437E-01	S0	-3.8547927E-01	S0	3.4328288E-01	S0		

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-142.6832581	-18.5728302	7.3714848
P (PER R/S)	.0000000	-3.7930794	-.1390026
R (PER R/S)	.0000000	1.4291840	-.1217285
AILERON (PER RAD)	-29.8520050	93.1127777	6.2604446
RUDDER (PER RAD)	31.1628113	13.6843510	-4.1703892

B Matrix:

A Matrix:

	P	R	PHI	AIL	RUD
BETA					
-2.556E+00	.6574E-01	-.9978E+00	-.5751E-01	-.5347E-01	.5582E-01
-.1857E+02	-.3793E+01	.1429E+01	.0000E+00	.9311E+02	.1368E+02
.7371E+01	-.1390E+00	-.1217E+00	.0000E+00	.6260E+01	-.4170E+01
.0000E+00	.1000E+01	.6588E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-2.4800354E-01	,+J(2.9022770E+00)
-2.4800354E-01	,+J(-2.9022770E+00)
-3.6932421E+00	,+J(0.0000000E+00)
1.8877365E-02	,+J(0.0000000E+00)

DYNAMICS

PERIOD=	2.1649	WD=	2.90228
ZETA=	.08514	WN=	2.91286

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
4.1703854E+00 S3
1.0237403E+01 S2
3.1141190E+01 S1
-5.9156340E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-5.3471737E-02 S3	9.3112778E+01 S3	6.2604446E+00 S3	0.0000000E+00 S3
-3.3519650E-01 S2	4.5072449E+01 S2	1.2009311E+01 S2	9.3525208E+01 S2
-4.1038227E+00 S1	8.0565845E+02 S1	5.3892731E+01 S1	4.5963632E+01 S1
1.2072630E+00 S0	-3.0424337E+00 S0	4.6157516E+01 S0	8.0920898E+02 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
5.5819687E-02 S3	1.3684351E+01 S3	-4.1703892E+00 S3	0.0000000E+00 S3
5.2794724E+00 S2	-1.8337955E+00 S2	-1.8375168E+01 S2	1.3409604E+01 S2
1.8208130E+01 S1	2.2731705E+01 S1	-1.2847195E+00 S1	-3.0443583E+00 S1
-3.1410569E-01 S0	-8.8885069E-02 S0	1.3467197E+00 S0	2.2647064E+01 S0

NUMERATOR AY/ AIL	NUMERATOR AY/ RUD
-2.9787445E+01 S4	3.1093413E+01 S4
-1.1645752E+02 S3	1.1562158E+02 S3
-2.5282568E+02 S2	-4.562554E+02 S2
-3.4551172E+02 S1	-1.6271694E+03 S1
-1.5459692E+02 S0	2.6407089E+01 S0

AIRCRAFT STATE

MACH = .70000 ALT = 100.00 VTRFPS = 781.59
 NZ = 1.00066 ALPHA = 2.72150 BETA = .00000
 GAMMA = .00000 THETA = 2.72151 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IXZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.56123 FLAPS = -.80706 STRAKES= -4.68337
 AILERON= .00000 RUDDER = .00000
 THRUST = 4077.173 X RPM = 21.68181

DYNAMICS

CXAERO = -.02479227 CYAERO = .00000000 CZAERO = -.11874694
 CLMAERO= .00000000 CMMAERO= .00362189 CNMAERO= .00000000
 VC KTS = 462.285 DYN PR = 723.88159 RHO = .00236997

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2655108	-5.4919415	1.7659721	.0000000
PITCH RATE(PER RAD)	.0000000	-7.2726564	-7.8932467	.0000000
STRAKE (PER RAD)	.0235242	-.1774627	-.2696343	.0000000
MACH (PER M #)	.0070645	-.0414199	-.0201787	326.3793945
ALPHA DOT (PER RAD)	.0000000	-.4739501	-1.1184864	.0000000
CANARD (PER RAD)	-.0017611	-.3626418	.9669358	.0000000
FLAPERON (PER RAD)	.0358100	-1.5469913	-.5789520	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0359478	.2022488	-.0528047
ROLL RATE (PER RAD)	.0000000	-.0047338	-.4276027
YAW RATE (PER RAD)	.0000000	-.1779084	.1357201
AILERON (PER RAD)	-.2177042	.0771123	.1940505
RUDDER (PER RAD)	.2079746	-.1221348	.0356349

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-1486.2680664	34.4691772	71.1161194
ALPHADOT (PER R/S)	-.5910622	-.1008528	.0000000
B (PER R/S)	-9.0902090	-.7133352	.0000000
VEL (PER FT/S)	-.0018786	-.0022763	-.0192128
STRAKE (PER RAD)	-48.0239563	-5.2757368	6.3659945
FLAPRN (PER RAD)	-418.6379395	-11.3279285	9.6906929
CANARD (PER RAD)	-98.1360779	18.9193268	-.4765867

B Matrix:

A Matrix:

VEL	ALPHA	B	THETA	STRAKE	FLAPRN	CANARD
-.1921E-01	.7112E+02	-.3711E+02	-.3214E+02	.6366E+01	.9691E+01	-.4766E+00
-.2404E-05	-.1902E+01	.9876E+00	-.1955E-02	-.6147E-01	-.5358E+00	-.1256E+00
-.2276E-02	-.3466E+02	-.8129E+00	.1972E-03	-.5270E+01	-.1127E+02	.1893E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-7.2410278E+00	+JC	0.0000000E+00
4.5263805E+00	+JC	0.0000000E+00
-9.8985508E-03	+JC	6.4784050E-02
-9.8985508E-03	+JC	-6.4784050E-02

DYNAMICS

PERIOD= 96.9866 WD= .06478
ZETA= .15104 WN= .06554

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
2.7344608E+00 S3
-3.2717407E+01 S2
-6.3749021E-01 S1
-1.4075744E-01 S0

NUMERATOR THT/STRK
0.0000000E+00 S3
-5.2695379E+00 S2
-1.2270535E+01 S1
-2.5253677E-01 S0

NUMERATOR Q/STRK
-5.2695379E+00 S3
-1.2270535E+01 S2
-2.5253677E-01 S1
5.4216143E-05 S0

NUMERATOR ALP/STRK
-6.1467018E-02 S3
-5.2554083E+00 S2
-1.0022920E-01 S1
4.3434203E-03 S0

NUMERATOR U/STRK
6.3659945E+00 S3
2.0846909E+02 S2
3.8688110E+01 S1
3.9178906E+02 S0

NUMERATOR THT/FLAP
0.0000000E+00 S3
-1.1273890E+01 S2
-4.0257339E+01 S1
-7.2674704E-01 S0

NUMERATOR Q/FLAP
-1.1273890E+01 S3
-4.0257339E+01 S2
-7.2674704E-01 S1
1.9094681E-04 S0

NUMERATOR ALP/FLAP
-5.3582478E-01 S3
-1.1580105E+01 S2
-1.7767787E-01 S1
3.8851853E-02 S0

NUMERATOR U/FLAP
9.6906929E+00 S3
4.0658618E+02 S2
7.0788794E+02 S1
1.2883340E+03 S0

NUMERATOR THT/CAN
0.0000000E+00 S3
1.8931992E+01 S2
3.2025665E+01 S1
6.3384253E-01 S0

NUMERATOR Q/CAN
1.8931992E+01 S3
3.2025665E+01 S2
6.3384253E-01 S1
-1.3016749E-04 S0

NUMERATOR ALP/CAN
-1.2560672E-01 S3
1.8592880E+01 S2
3.3363402E-01 S1
9.8378770E-03 S0

NUMERATOR U/CAN
-4.7658670E-01 S3
-7.1280255E+02 S2
-4.4537817E+02 S1
-1.0201667E+03 S0

NUMERATOR AN/CAN
-9.8061829E+01 S4
-2.6474219E+02 S3
-2.4713215E+04 S2
-4.3823975E+02 S1
1.0699129E+00 S0

NUMERATOR AN/FLAP
-4.1832129E+02 S4
-2.3906250E+02 S3
3.1273184E+04 S2
5.3620776E+02 S1
-1.2592897E+00 S0

NUMERATOR AN/STRK
-4.7987625E+01 S4
1.1031250E+01 S3
9.4933711E+03 S2
1.8180263E+02 S1
-4.2811507E-01 S0

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-280.3422852	-37.1875916	12.7345037
P (PER R/S)	.0000000	-6.0475683	-.2160444
R (PER R/S)	.0000000	1.9312702	-.1510251
AILERON (PER RAD)	-58.9138794	159.8407135	10.9076433
RUDDER (PER RAD)	56.2808990	25.5013428	-7.5842838

A Matrix:

B Matrix:

BETA	P	R	PHI	AIL	RUO
-.3587E+00	.4748E-01	-.9989E+00	.4112E-01	-.7538E-01	.7201E-01
-.3719E+02	-.6048E+01	.1831E+01	.0000E+00	.1598E+03	.2550E+02
.1273E+02	-.2160E+00	-.1510E+00	.0000E+00	.1091E+02	-.7584E+01
.0000E+00	.1000E+01	.4753E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-3.0316657E-01 +J(-3.7920914E+00)
 -3.0316657E-01 +J(-3.7920914E+00)
 -5.9613857E+00 +J(0.0000000E+00)
 1.0364234E-02 +J(0.0000000E+00)

DYNAMICS

PERIOD= 1.6569 WD= 3.79209
 ZETA= .07969 WN= 3.80419

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
6.557276E+00 S3
1.8018143E+01 S2
8.6084122E+01 S1
-8.9422387E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-7.5377285E-02 S3	1.5984071E+02 S3	1.0907643E+01 S3	0.000000E+00 S3
-3.7731848E+00 S2	1.0425009E+02 S2	3.4384506E+01 S2	1.6035919E+02 S2
-2.2806900E+01 S1	2.4528560E+03 S1	1.2077025E+02 S1	1.0588454E+02 S1
1.8753843E+00 S0	-4.7717667E+00 S0	1.0037524E+02 S0	2.4585969E+03 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
7.2008550E-02 S3	2.5501343E+01 S3	-7.5842838E+00 S3	0.000000E+00 S3
9.232902E+00 S2	-3.5684490E+00 S2	-5.3179276E+01 S2	2.5140823E+01 S2
5.1969360E+01 S1	4.0332016E+01 S1	-1.0275844E+01 S1	-6.0963030E+00 S1
-5.1317596E-01 S0	-8.3435714E-02 S0	1.7561102E+00 S0	3.9843552E+01 S0

NUMERATOR AY/ AIL	NUMERATOR AY/ RUD
-5.8847443E+01 S4	5.6217453E+01 S4
-3.6185547E+02 S3	3.4072266E+02 S3
1.6349121E+01 S2	-1.6201277E+03 S2
1.3205625E+03 S1	-9.7238672E+03 S1
-4.7310498E+02 S0	9.3620514E+01 S0

AIRCRAFT STATE

MACH = .90000 ALT = 100.00 VTRFPS = 1004.90
 NZ = .99818 ALPHA = 2.53971 BETA = .00000
 GAMMA = .00000 THETA = 2.53972 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 Ixz = 1827.00 XCG = 450.56006 ZCG = 64.89000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.79026 FLAPS = -3.01641 STRAKES= -4.97670
 AILERON= .00000 RUDDER = .00000
 THRUST = 7406.086 X RPM = 43.20621

DYNAMICS

CXAERO = -.03025946 CYAERO = .00000000 CZAERO = -.07184625
 CLMAERO= .00000000 CMMAERO= .00399078 CNMAERO= .00000000
 VC KTS = 594.425 DYN PR = 1196.62061 RHO = .00236997

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2901374	-7.2770729	2.0986538	.0000000
PITCH RATE(PER RAD)	.0000000	-8.0015907	-8.8425236	.0000000
STRAKE (PER RAD)	.0086705	-.1324789	-.1864818	.0000000
MACH (PER M #)	-.0719362	-.3839715	-.0491222	-7836.8359375
ALPHA DOT (PER RAD)	.0000000	-.4487051	-1.2793922	.0000000
CANARD (PER RAD)	-.0184006	-.5848925	1.2607012	.0000000
FLAPERON (PER RAD)	.0309399	-1.5183439	-.6590108	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0203457	.1804873	-.0680822
ROLL RATE (PER RAD)	.0000000	.0013629	-.5398493
YAW RATE (PER RAD)	.0000000	-.1513517	.1414860
AILERON (PER RAD)	-.2980995	.0500296	.1285638
RUDDER (PER RAD)	.1689356	-.1009741	.0310236

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	FT/SEC**2	M	RAD/SEC**2	X	FT/SEC**2
ALPHA (PER RAD)	-3258.6384277		67.6941833		127.0552521	
ALPHADOT (PER R/S)	-7196688		-1.1483653		.0000000	
Q (PER R/S)	-12.8588362		-1.0274448		.0000000	
VEL (PER FT/S)	-0740439		-0.040359		-.0756097	
STRAKE (PER RAD)	-59.2633972		-6.0316086		3.8786955	
FLAPRN (PER RAD)	-679.2189941		-21.3152008		13.8407316	
CANARD (PER RAD)	-261.6472168		40.7764130		-8.2313652	

B Matrix:

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.7561E-01	.1271E+03	-.4453E+02	-.3214E+02	.3879E+01	.1384E+02	-.8231E+01
-.7370E-04	-.3244E+01	.9865E+00	-.1419E-02	-.5899E-01	-.6761E+00	-.2604E+00
-.4025E-02	-.6818E+02	-.1174E+01	.2105E-03	-.6023E+01	-.2121E+02	.4082E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

6.0714560E+00	+J(0.0000000E+00)
-1.0483432E+01	+J(0.0000000E+00)
-4.0539443E-02	+J(8.5936368E-02)
-4.0539443E-02	+J(-8.5936368E-02)

DYNAMICS

PERIOD= 73.1144	WD= .08594
ZETA= .42665	WN= .09502

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
4.4930344E+00 S3
-6.3282684E+01 S2
-5.1201115E+00 S1
-5.7420695E-01 S0

NUMERATOR U/STRK
3.8786955E+00 S3
2.7782764E+02 S2
2.3279019E+02 S1
7.5865259E+02 S0

NUMERATOR U/FLAP
1.3840732E+01 S3
9.1990723E+02 S2
2.160439E+03 S1
3.6985046E+03 S0

NUMERATOR U/CAN
-3.2313652E+00 S3
-1.8868857E+03 S2
-8.1723340E+02 S1
-3.69226934E+03 S0

NUMERATOR AN/STRK
-5.9220947E+01 S4
7.4453125E+00 S3
2.3641250E+04 S2
1.8446211E+03 S1
-2.8205986E+00 S0

NUMERATOR ALP/STRK
-5.9990270E-02 S3
-6.0154419E+00 S2
-4.7083360E-01 S1
-5.8970228E-03 S0

NUMERATOR ALP/FLAP
-6.7608869E-01 S3
-2.1773895E+01 S2
-1.6167126E+00 S1
3.9818518E-02 S0

NUMERATOR ALP/CAN
-2.6044136E-01 S3
3.9938599E+01 S2
3.1773291E+00 S1
1.2548280E-01 S0

NUMERATOR AN/FLAP
-6.7873242E+02 S4
-5.6119141E+02 S3
1.1536794E+05 S2
8.6990000E+03 S1
-1.3419224E+01 S0

NUMERATOR Q/STRK
-6.0228567E+00 S3
-2.4028549E+01 S2
-1.8774567E+00 S1
1.4338866E-04 S0

NUMERATOR Q/FLAP
-2.1214890E+01 S3
-1.1656543E+02 S2
-8.7908335E+00 S1
8.8287168E-04 S0

NUMERATOR Q/CAN
4.0815048E+01 S3
1.1775192E+02 S2
9.3310308E+00 S1
-4.3806969E-04 S0

NUMERATOR AN/CAN
-2.6145972E+02 S4
-8.7987891E+02 S3
-1.1496444E+05 S2
-9.0736719E+03 S1
1.3742829E+01 S0

NUMERATOR THT/STRK
0.0000000E+00 S3
-6.0229567E+00 S2
-2.4028549E+01 S1
-1.8774567E+00 S0

NUMERATOR THT/FLAP
0.0000000E+00 S3
-2.1214890E+01 S2
-1.1656543E+02 S1
-8.7908335E+00 S0

NUMERATOR THT/CAN
0.0000000E+00 S3
4.0815048E+01 S2
1.1775192E+02 S1
9.3310308E+00 S0

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-456.4433594	-83.0087433	17.8068695
P (PER R/S)	.0000000	-9.8118830	-.3391372
R (PER R/S)	.0000000	2.4760838	-.1487379
AILERON (PER RAD)	-133.3524323	175.0079193	11.8228521
RUDDER (PER RAD)	75.5719910	36.9504395	-10.2921162

B Matrix:

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.4542E+00	.4431E-01	-.9990E+00	.3199E-01	-.1327E+00	.7520E-01
-.8301E+02	-.9812E+01	.2476E+01	.0000E+00	.1750E+03	.3695E+02
-.1781E+02	-.3391E+00	-.1487E+00	.0000E+00	.1182E+02	-.1029E+02
.0000E+00	.1000E+01	.4436E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-9.6807079E+00 ,+J(0.0000000E+00)
 -3.7025166E-01 ,+J(4.5882130E+00)
 -3.7025166E-01 ,+J(-4.5882130E+00)
 6.3537657E-03 ,+J(0.0000000E+00)

DYNAMICS

PERIOD= 1.3694 WD= 4.58821
 ZETA= .08043 WN= 4.60313

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
1.0414840E+01 S3
2.8291046E+01 S2
2.0493846E+02 S1
-1.3031721E+00 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-1.3270253E-01 S3	1.7500792E+02 S3	1.1822852E+01 S3	0.000000E+00 S3
-5.3781748E+00 S2	1.4591204E+02 S2	5.9659892E+01 S2	1.7553232E+02 S2
-4.8837036E+01 S1	4.1146211E+03 S1	1.8038941E+02 S1	1.4845827E+02 S1
1.8493738E+00 S0	-5.8051500E+00 S0	1.3106953E+02 S0	4.1226250E+03 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
7.5203717E-02 S3	3.6950439E+01 S3	-1.0292116E+01 S3	0.000000E+00 S3
1.2668412E+01 S2	-9.4471903E+00 S2	-1.1685205E+02 S2	3.6493927E+01 S2
1.1385928E+02 S1	-2.0286311E+02 S1	-4.5003753E+01 S1	-1.4630171E+01 S1
-8.0048782E-01 S0	2.8070343E-01 S0	-6.2804480E+00 S0	-2.0485934E+02 S0

NUMERATOR AY/ AIL	NUMERATOR AY/ RUD
-1.3322145E+02 S4	7.5497787E+01 S4
-1.3229648E+03 S3	7.4023755E+02 S3
-1.2696523E+03 S2	-3.7568306E+03 S2
-5.0393750E+03 S1	-3.6481918E+04 S1
-6.7044287E+02 S0	2.6716089E+02 S0

04/10/92 08:58:37 TASK # 03000006

MODELS ENCORE COMPUTER CORP. MPX-32 3.5

AIRCRAFT STATE

MACH = .95000 ALT = 100.00 VTRFPS = 1060.72
 NZ = .99879 ALPHA = 2.38942 PETA = .00000
 GAMMA = .00000 THETA = 2.38943 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IXZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.84744 FLAPS = -3.81803 STRAKES= -4.90025
 AILERON= .00000 RUDDER = .00000
 THRUST = 10094.000 X RPM = 57.11710

DYNAMICS

CXAERO = -.03322172 CYAERO = .00000000 CZAERO = -.06449628
 CLMAERO= .00000000 CMMAERO= .00487923 CNMAERO= .00000000
 VC KTS = 627.465 DYN PR = 1333.27148 RHO = .00236997

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2862445	-7.7614584	1.7732210	.0000000
PITCH RATE(PER RAD)	.0000000	-8.0908966	-9.0930719	.0000000
STRAKE (PER RAD)	.0080152	-.1237546	-.1820582	.0000000
MACH (PER M #)	-.1556872	-.4183871	-.0366994	-7522.5390625
ALPHA DOT (PER RAD)	.0000000	-.4385120	-1.3459539	.0000000
CANARD (PER RAD)	-.0169569	-.6578407	1.1119719	.0000000
FLAPERON (PER RAD)	.0301778	-1.1464062	-.6421686	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0196772	.1633657	-.0727537
ROLL RATE (PER RAD)	.0000000	.0134417	-.5889903
YAW RATE (PER RAD)	.0000000	-.1318330	.1582530
AILERON (PER RAD)	-.3320057	.0541682	.1064506
RUDDER (PER RAD)	.1457419	-.0964976	.0259046

LONGITUDINAL SYSTEM
 BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-3872.7272949	63.7543182	137.7661591
ALPHADOT (PER R/S)	-7425607	-1647926	-0.0000000
Q (PER R/S)	-13.7247009	-1.1152544	-0.0000000
VEL (PER FT/S)	-0951165	-0032466	-1245458
STRAKE (PER RAD)	-61.6826477	-6.5609875	3.9949694
FLAPRN (PER RAD)	-571.3999023	-23.1423798	15.0414438
CANARD (PER RAD)	-327.8857422	40.0730591	-8.4517632

A Matrix:

S Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-1245E+00	1378E+03	-4422E+02	-3215E+02	3995E+01	1504E+02	-8452E+01
-8969E-04	-3632E+01	9864E+00	-1265E-02	-5816E-01	-5388E+00	-3092E+00
-3232E-02	-6436E+02	-1278E+01	-2084E-03	-6551E+01	-2305E+02	4012E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.0526440E+01	+J(-	0.0000000E+00)
5.6032991E+00	+J(-	0.0000000E+00)
-6.5414846E-02	+J(-	7.1679235E-02)
-6.5414846E-02	+J(-	-7.1679235E-02)

DYNAMICS

PERIOD= 87.6570	WD= .07168
ZETA= .67409	WN= .09704

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
5.0539827E+00 S3
-5.8328964E+01 S2
-7.6702490E+00 S1
-5.5542374E-01 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
3.9949694E+00 S3		-5.8161277E-02 S3		-6.5514030E+00 S3		0.000000E+00 S3	
3.0140063E+02 S2		-6.5439548E+00 S2		-2.8495255E+01 S2		-6.5514030E+00 S2	
2.9364063E+02 S1		-8.3663672E-01 S1		-3.5709648E+00 S1		-2.8495255E+01 S1	
8.9082886E+02 S0		-1.1789785E-02 S0		1.0344552E-04 S0		-3.5709648E+00 S0	
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
1.5041444E+01 S3		-5.3877956E-01 S3		-2.3053589E+01 S3		0.000000E+00 S3	
1.0194111E+03 S2		-2.3496017E+01 S2		-1.2177695E+02 S2		-2.3053589E+01 S2	
1.8551101E+03 S1		-2.9526215E+00 S1		-1.5112339E+01 S1		-1.2177695E+02 S1	
3.8260251E+03 S0		-6.7544021E-03 S0		3.9149029E-04 S0		-1.5112339E+01 S0	
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-8.4517632E+00 S3		-3.0916721E-01 S3		4.0124008E+01 S3		0.000000E+00 S3	
-1.8586455E+03 S2		3.9143860E+01 S2		1.3164616E+02 S2		4.0124008E+01 S2	
-9.9440063E+02 S1		5.0604296E+00 S1		1.6552002E+01 S1		1.3164616E+02 S1	
-4.0780413E+03 S0		1.4141864E-01 S0		-6.0487841E-04 S0		1.6552002E+01 S0	
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-6.1639465E+01 S4		-5.7100000E+02 S4		-3.2765625E+02 S4			
7.8906250E+00 S3		-4.688672E+02 S3		-1.0387617E+03 S3			
2.9303883E+04 S2		1.2589938E+05 S2		-1.3410206E+05 S2			
3.7338008E+03 S1		1.5845586E+04 S1		-1.7213391E+04 S1			
-4.9017200E+00 S0		-2.0686066E+01 S0		2.2843338E+01 S0			

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y	L	N
	FT/SEC**2	RAD/SEC**2	RAD/SEC**2
BETA (PER RAD)	-508.2351074	-100.3700409	17.3789825
P (PER R/S)	.0000000	-11.2917381	-.3707060
R (PER R/S)	.0000000	2.9483719	-.1133894
AILERON (PER RAD)	-165.4807892	162.1175232	12.5582976
RUDDER (PER RAD)	72.6417542	34.2631989	-9.8583164

A Matrix:

B Matrix:

BETA	P	R	PHI	AIL	RUD
-.4791E+00	.4169E-01	-.9991E+00	.3031E-01	-.1560E+00	-.6848E-01
-.1004E+03	-.1129E+02	.2948E+01	.0000E+00	.1621E+03	.3426E+02
.1738E+02	-.3707E+00	-.1134E+00	.0000E+00	.1256E+02	-.9858E+01
.0000E+00	.1000E+01	.4173E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.1146408E+01 +J(0.0000000E+00)
-3.7210846E-01 +J(4.5856256E+00)
-3.7210846E-01 +J(-4.5856256E+00)
6.3711554E-03 +J(0.0000000E+00)

DYNAMICS

PERIOD= 1.3702 WD= 4.58563
ZETA= .08088 WN= 4.60070

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
1.1884316E+01 S3
2.9386398E+01 S2
2.3573973E+02 S1
-1.5031719E+00 S0

NUMERATOR PHI/ AIL
0.0000000E+00 S3
1.6264154E+02 S2
1.5229156E+02 S1
4.1019023E+03 S0

NUMERATOR R/ AIL
1.2558288E+01 S3
8.5013489E+01 S2
1.7274182E+02 S1
1.2358406E+02 S0

NUMERATOR PHI/ RUD
0.0000000E+00 S3
3.3851837E+01 S2
-2.0960175E+01 S1
-4.0551123E+02 S0

NUMERATOR R/ RUD
-9.8583164E+00 S3
-1.2755293E+02 S2
-5.9862381E+01 S1
-1.1940517E+01 S0

NUMERATOR /

NUMERATOR P/ AIL
1.6211752E+02 S3
1.4874419E+02 S2
4.0946960E+03 S1
-5.1479092E+00 S0

NUMERATOR P/ RUD
3.4263199E+01 S3
-1.5637726E+01 S2
-4.0301343E+02 S1
5.0130177E-01 S0

NUMERATOR AY/ RUD
7.2578644E+01 S4
8.1737231E+02 S3
-4.1087070E+03 S2
-4.5921355E+04 S1
3.5881543E+02 S0

NUMERATOR BETA/ AIL
-1.5600729E-01 S3
-7.5678205E+00 S2
-7.4767776E+01 S1
1.7826347E+00 S0

NUMERATOR BETA/ RUD
6.8483174E-02 S3
1.2059273E+01 S2
1.2405035E+02 S1
-9.2013830E-01 S0

NUMERATOR AY/ AIL
-1.6533691E+02 S4
-1.8803555E+03 S3
-9.4767578E+02 S2
-1.0124375E+03 S1
-6.5753320E+02 S0

AIRCRAFT STATE

MACH = .50000 ALT = 10000.00 VTRFPS = 538.94
 NZ = .99925 ALPHA = 4.38708 BETA = .00000
 GAMMA = .00000 THETA = 4.38710 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IYZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS = -3.04494 FLAPS = 3.28307 STRAKES = -5.49912
 AILERON = .00000 RUDDER = .00000
 THRUST = 2003.690 X RPM = 13.87500

DYNAMICS

CXAERO = -.01664839 CYAERO = .000000000 CIAERO = -.33655757
 CLMAERO = .000000000 CMMAERO = .00505804 CNMAERO = .000000000
 VC KTS = 276.854 DYN PR = 254.96179 RHO = .00175561

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.3709802	-4.7869196	1.5520849	.0000000
PITCH RATE(PER RAD)	.0000000	-6.7314140	-7.4655275	.0000000
STRAKE (PER RAD)	.0315772	-.2178361	-.3278557	.0000000
MACH (PER M #)	.0026060	.0076967	.0336312	356.5844727
ALPHA DOT (PER RAD)	.0000000	-.4772125	-1.0246563	.0000000
CANARD (PER RAD)	-.0129858	-.2147608	.9620700	.0000000
FLAPERON (PER RAD)	.0381018	-1.4610481	-.5628068	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0611315	.2462735	-.0490339
ROLL RATE (PER RAD)	.0000000	-.0144992	-.3667075
YAW RATE (PER RAD)	.0000000	-.2016139	.1600633
AILERON (PER RAD)	-.2164459	.0844897	.2238584
RUDDER (PER RAD)	.2375308	-.1394036	.0398873

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-458.4580078	10.6478081	34.9202728
ALPHADOT (PER R/S)	-3028936	-0.0470233	.0000000
Q (PER R/S)	-4.3295946	-3.446231	.0000000
VEL (PER FT/S)	-0.0532582	-0.0011805	-0.0099922
STRAKE (PER RAD)	-20.7629089	-2.2594261	3.0097570
FLAPRN (PER RAD)	-139.2589264	-3.8785982	3.6316519
CANARD (PER RAD)	-20.4698029	5.9409790	-1.2377377

A Matrix:

B Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.9992E-02	.3492E+02	-.4123E+02	-.3208E+02	.3010E+01	.3632E+01	-.1238E+01
-.9906E-04	-.8527E+00	.9914E+00	-.4577E-02	-.3862E-01	-.2590E+00	-.3807E-01
-.1176E-02	.1069E+02	-.3912E+00	.2152E-03	-.2258E+01	-.3846E+01	.5943E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

2.6514835E+00 ,+J(0.0000000E+00)
 -3.8916149E+00 ,+J(0.0000000E+00)
 -6.8945885E-03 ,+J(7.9574466E-02)
 -6.8945885E-03 ,+J(-7.9574466E-02)

DYNAMICS

PERIOD= 78.9598 WD= .07957
 ZETA= .08632 WN= .07987

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 1.2539215E+00 S3
 -1.0295012E+01 S2
 -1.3444299E-01 S1
 -6.5829039E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
3.0097570E+00	S3	-3.8617037E-02	S3	-2.2576103E+00	S3	0.0000000E+00	S3
9.5466263E+01	S2	-2.2539511E+00	S2	-2.3638706E+00	S2	-2.2576103E+00	S2
5.9226990E+01	S1	-2.3144443E-02	S1	-3.5786062E-02	S1	-2.3638706E+00	S1
7.5502930E+01	S0	-5.5972412E-03	S0	2.1645419E-06	S0	-3.5786062E-02	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
3.6316519E+00	S3	-2.5900835E-01	S3	-3.8664188E+00	S3	0.0000000E+00	S3
1.5486748E+02	S2	-3.9373884E+00	S2	-6.1080103E+00	S2	-3.8664188E+00	S2
1.9940871E+02	S1	-2.9165860E-02	S1	-7.0826054E-02	S1	-6.1080103E+00	S1
1.9336386E+02	S0	-2.3180142E-03	S0	4.0095174E-06	S0	-7.0826054E-02	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-1.2377377E+00	S3	-3.8071889E-02	S3	5.9427691E+00	S3	0.0000000E+00	S3
-2.4786211E+02	S2	5.8764114E+00	S2	4.7212553E+00	S2	5.9427691E+00	S2
-1.6485291E+02	S1	5.9128899E-02	S1	7.1236491E-02	S1	4.7212553E+00	S1
-1.5051492E+02	S0	2.0041022E-02	S0	-5.4730081E-06	S0	7.1236491E-02	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-2.0751205E+01	S4	-1.3918044E+02	S4	-2.0458267E+01	S4		
1.9663086E+00	S3	-3.8136230E+01	S3	-3.5657715E+01	S3		
1.2522539E+03	S2	3.2570054E+03	S2	-2.4906094E+03	S2		
1.0404501E+01	S1	2.1780914E+01	S1	-1.5890806E+01	S1		
-8.9236557E-02	S0	-1.7646515E-01	S0	1.7826176E-01	S0		

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-101.1411743	-11.5807762	5.6135473
P (PER R/S)	.0000000	-2.6517696	-.1011655
R (PER R/S)	.0000000	1.1052389	-.0858497
AILERON (PER RAD)	-20.6304169	64.9017029	4.3212833
RUDDER (PER RAD)	22.6401215	10.0269642	-3.0568190

B Matrix:

A Matrix:

	P	R	PHI	AIL	RUD
BETA					
-1.1877E+00	.7649E-01	-.9971E+00	.5952E-01	-.3828E-01	.4201E-01
-1.1158E+02	-.2652E+01	.1105E+01	.0000E+00	.6490E+02	.1003E+02
.5614E+01	-.1012E+00	-.8585E-01	.0000E+00	.4321E+01	-.3037E+01
.0000E+00	.1000E+01	.7672E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.9003361E-01 ,+J(2.5283260E+00)
-1.9003361E-01 ,+J(-2.5283260E+00)
-2.5684462E+00 ,+J(0.0000000E+00)
2.3223985E-02 ,+J(0.0000000E+00)

DYNAMICS

PERIOD= 2.4851 WD= 2.52833
ZETA= .07495 WN= 2.53546

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
2.9252863E+00 S3
7.3361835E+00 S2
1.6339218E+01 S1
-3.8346112E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-3.8279757E-02 S3	6.4901703E+01 S3	4.3212833E+00 S3	0.000000E+00 S3
5.5116093E-01 S2	2.2971054E+01 S2	5.4893131E+00 S2	6.5233215E+01 S2
-2.1740794E-01 S1	4.1490088E+02 S1	3.2000549E+01 S1	2.3392181E+01 S1
6.3829064E-01 S0	-1.8927374E+00 S0	2.4665054E+01 S0	4.1735596E+02 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
4.2008765E-02 S3	1.0026964E+01 S3	-3.0568190E+00 S3	0.000000E+00 S3
3.9298658E+00 S2	-1.1224680E+00 S2	-9.4582081E+00 S2	9.7924490E+00 S2
9.4981928E+00 S1	2.0571686E+01 S1	5.6064719E-01 S1	-1.8480883E+00 S1
-1.9151574E-01 S0	-9.5429540E-02 S0	1.2432508E+00 S0	2.0614700E+01 S0

NUMERATOR AY/ AIL	NUMERATOR AY/ RUD
-2.0569962E+01 S4	2.2573776E+01 S4
-5.7346680E+01 S3	5.5773926E+01 S3
-2.0674951E+02 S2	-2.4638206E+02 S2
-3.1608984E+02 S1	-5.9043335E+02 S1
-5.6650848E+01 S0	1.0691689E+01 S0

AIRCRAFT STATE

MACH = .70000 ALT = 10000.00 VTRFPS = 754.51
 NZ = .99898 ALPHA = 3.08648 BETA = .00000
 GAMMA = .00000 THETA = 3.08650 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 Ixz = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.60869 FLAPS = -.05130 STRAKES= -3.79884
 AILERON= .00000 RUDDER = .00000
 THRUST = 2870.434 X RPM = 19.01076

DYNAMICS

CXAERO = -.02176704 CYAERO = .00000000 CZAERO = -.17198908
 CLMAERO= .00000000 CMMAERO= .00370841 CNMAERO= .00000000
 VC KTS = 390.706 DYN PR = 499.72461 RHO = .00175561

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2805251	-5.2550144	1.7002331	.0000000
PITCH RATE(PER RAD)	.0000000	-7.2726946	-7.8930044	.0000000
STRAKE (PER RAD)	.0223656	-.1955577	-.2842008	.0000000
MACH (PER M #)	.0050186	-.0621646	.0104233	762.3315430
ALPHA DOT (PER RAD)	.0000000	-.4739495	-1.1184864	.0000000
CANARD (PER RAD)	-.0030734	-.3219798	.9120272	.0000000
FLAPERON (PER RAD)	.0353100	-1.5469923	-.5789520	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0534906	.2154524	-.0538382
ROLL RATE (PER RAD)	.0000000	-.0087045	-.3996911
YAW RATE (PER RAD)	.0000000	-.1813709	.1437543
AILERON (PER RAD)	-.2178183	.0770122	.1979908
RUDDER (PER RAD)	.2276835	-.1338190	.0390402

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-982.7670898	22.0104767	51.8526764
ALPHADOT (PER R/S)	-4224018	-0720745	.0000000
Q (PER R/S)	-6.5005484	-5100984	.0000000
VEL (PER FT/S)	-0257472	-0013882	-0122058
STRAKE (PER RAD)	-36.5332794	-3.8388081	4.1782598
FLAPRN (PER RAD)	-289.0029297	-7.8201237	6.6898737
CANARD (PER RAD)	-60.1509552	12.3190975	-5741571

B Matrix:

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-0.1221E-01	.5185E+02	-0.4063E+02	-0.3213E+02	.4178E+01	.6690E+01	-0.5742E+00
-0.3415E-04	-0.1304E+01	.9903E+00	-0.2298E-02	-0.4846E-01	-0.3834E+00	-0.7979E-01
-0.1386E-02	.2300E+02	-0.5815E+00	.1456E-03	-0.3535E+01	-0.7792E+01	.1232E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-5.7359972E+00 +J(0.0000000E+00)
3.8522091E+00 +J(0.0000000E+00)
-6.8098679E-03 +J(6.0807392E-02)
-6.8098679E-03 +J(-6.0807392E-02)

DYNAMICS

PERIOD=103.3293 WD= .06081
ZETA= .11130 WN= .06119

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 1.8973961E+00 S3
 -2.2066772E+01 S2
 -2.9392242E-01 S1
 -8.2635760E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
4.1782598E+00	S3	-4.8462857E-02	S3	-3.9353148E+00	S3	0.0000000E+00	S3
1.6117412E+02	S2	-3.9290100E+00	S2	-6.1674843E+00	S2	-3.8353148E+00	S2
8.1061066E+01	S1	-4.6317209E-02	S1	-8.8720918E-02	S1	-6.1674843E+00	S1
1.9712996E+02	S0	-1.9746483E-03	S0	2.4686960E-05	S0	-8.8720918E-02	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
6.6898737E+00	S3	-3.8337398E-01	S3	-7.7924919E+00	S3	0.0000000E+00	S3
3.0930371E+02	S2	-7.9487743E+00	S2	-1.9082611E+01	S2	-7.7924919E+00	S2
4.6202100E+02	S1	-7.7531457E-02	S1	-2.3512195E-01	S1	-1.9082611E+01	S1
6.1099902E+02	S0	8.8677257E-03	S0	1.4612412E-04	S0	-2.3512195E-01	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-5.7415712E-01	S3	-7.9792678E-02	S3	1.2324848E+01	S3	0.0000000E+00	S3
-5.0591870E+02	S2	1.2164309E+01	S2	1.4383293E+01	S2	1.2324848E+01	S2
-3.3069116E+02	S1	1.4256889E-01	S1	2.0257962E-01	S1	1.4383293E+01	S1
-4.5872925E+02	S0	1.6549437E-02	S0	2.9924684E-07	S0	2.0257962E-01	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-3.6512817E+01	S4	-2.8884106E+02	S4	-6.0117279E+01	S4		
4.7500000E+00	S3	-1.1774609E+02	S3	-1.2095313E+02	S3		
4.6051563E+03	S2	1.4305277E+04	S2	-1.0707871E+04	S2		
5.4747391E+01	S1	1.5076866E+02	S1	-1.1524176E+02	S1		
-1.7229456E-01	S0	-5.1740390E-01	S0	3.5071158E-01	S0		

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y	L	N
	FT/SEC**2	RAD/SEC**2	RAD/SEC**2
BETA (PER RAD)	-196.8089447	-25.9942017	9.4121571
P (PER R/S)	.0000000	-4.0439014	-1481572
R (PER R/S)	.0000000	1.3895693	-1082057
AILERON (PER RAD)	-40.6919403	112.5526428	7.6019907
RUDDER (PER RAD)	42.5349426	19.2866974	-5.7366943

A Matrix:

BETA	P	R	PHI	AIL	RUD
-2.2608E+00	.5384E-01	-.9985E+00	.4258E-01	-.5393E-01	.5637E-01
-.2599E+02	-.4044E+01	.1390E+01	.0000E+00	.1126E+03	.1929E+02
.9412E+01	-.1432E+00	-.1082E+00	.0000E+00	.7602E+01	-.5737E+01
.0000E+00	.1000E+01	.5392E-01	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-2.2881031E-01	+J(3.2728939E+00)
-2.2881031E-01	+J(-3.2728939E+00)
-3.9677877E+00	+J(0.0000000E+00)
1.2488514E-02	+J(0.0000000E+00)

DYNAMICS

PERIOD= 1.9198	WD= 3.27289
ZETA= .06974	WN= 3.28098

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
4.412949E+00 S3
1.252460E+01 S2
4.255258E+01 S1
-5.333766E-01 S0

NUMERATOR PHI/ AIL
0.000000E+00 S3
1.129625E+02 S2
5.434074E+01 S1
1.264251E+03 S0

NUMERATOR R/ AIL
7.601907E+00 S3
1.554152E+01 S2
6.908758E+01 S1
5.352206E+01 S0

NUMERATOR P/ AIL
1.125526E+02 S3
5.350273E+01 S2
1.260526E+03 S1
-2.886896E+00 S0

NUMERATOR BETA/ AIL
-5.393142E-02 S3
-1.754735E+00 S2
-8.046029E+00 S1
1.000670E+00 S0

NUMERATOR PHI/ RUD
0.000000E+00 S3
1.897735E+01 S2
-3.776263E+00 S1
3.126043E+01 S0

NUMERATOR R/ RUD
-5.736694E+00 S3
-2.702186E+01 S2
-2.688749E+00 S1
1.380014E+00 S0

NUMERATOR P/ RUD
1.928669E+01 S3
-2.319213E+00 S2
3.140541E+01 S1
-7.455462E-02 S0

NUMERATOR BETA/ RUD
5.637406E-02 S3
7.000897E+00 S2
2.654579E+01 S1
-3.103951E-01 S0

NUMERATOR /

NUMERATOR AY/ RUD
4.247325E+01 S4
1.689477E+02 S3
-8.741564E+02 S2
-3.414139E+03 S1
3.844236E+01 S0

NUMERATOR AY/ AIL
-4.063293E+01 S4
-1.670390E+02 S3
-1.554980E+02 S2
-1.491093E+02 S1
-1.752311E+02 S0

AIRCRAFT STATE

MACH = .90000 ALT = 10000.00 VTRFPS = 970.09
 NZ = .99633 ALPHA = 2.82944 BETA = .00000
 GAMMA = .00000 THETA = 2.82945 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IYZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AP MODE: F

CONTRCLS

CANARDS= -3.22420 FLAPS = -2.98074 STRAKES= -2.66410
 AILERON= .00000 RUDDER = .00000
 THRUST = 5079.176 X RPM = 32.59435

DYNAMICS

CXAERO = -.02808380 CYAERO = .000000000 CZAERO = -.10405451
 CLMAERO= .000000000 CMMAERO= .00395265 CNMAERO= .000000000
 VC KTS = 507.130 DYN PR = 826.07471 RHO = .00175561

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2997032	-6.6399498	1.9332123	.0000000
PITCH RATE(PER RAD)	.0000000	-8.0015631	-8.8423271	.0000000
STRAKE (PER RAD)	.0074209	-.1602557	-.1974170	.0000000
YACH (PER M #)	-.0734413	-.3259748	-.0521473	270.5078125
ALPHA DOT (PER RAD)	.0000000	-.4487630	-1.2793684	.0000000
CANARD (PER RAD)	-.0131636	-.4665840	1.1169062	.0000000
FLAPERON (PER PAD)	.0309398	-1.5183449	-.6590127	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0427599	.1939430	-.0714840
ROLL RATE (PER RAD)	.0000000	-.0018275	-.4675964
YAW RATE (PER RAD)	.0000000	-.1550213	.1507193
AILERON (PER RAD)	-.2978753	.0505683	.1375768
RUDDER (PER RAD)	.1948928	-.1155093	.0359199

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-2053.1762695	43.0176086	90.4667358
ALPHADOT (PER R/S)	-.5144659	-.1060455	.0000000
Q (PER R/S)	-9.1954689	-.7347212	.0000000
VEL (PER FT/S)	-.0551082	-.0030980	-.0430771
STRAKE (PER RAD)	-49.4898529	-4.4080257	2.2917156
FLAPRN (PER RAD)	-468.3925781	-14.7147722	9.5547733
CANARD (PER RAD)	-144.0895996	24.9389428	-4.0651712

A Matrix:

B Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.4308E-01	.9047E+02	-.4789E+02	-.3213E+02	.2292E+01	.9553E+01	-.4065E+01
-.5685E-04	-.2118E+01	.9900E+00	-.1638E-02	-.5105E-01	-.4837E+00	-.1486E+00
-.3092E-02	.4324E+02	-.8397E+00	.1737E-03	-.4403E+01	-.1466E+02	.2495E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

5.1095190E+00	+J(-0.0000000E+00)
-8.0625048E+00	+J(-0.0000000E+00)
-2.3866307E-02	+J(-8.0089688E-02)
-2.3866307E-02	+J(-8.0089688E-02)

DYNAMICS

PERIOD= 78.4519	WD= .08009
ZETA= .28558	WN= .08357

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
3.0007257E+00 S3
-4.1046341E+01 S2
-1.9464340E+00 S1
-2.8658420E-01 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
2.2917156E+00	S3	-5.1051043E-02	S3	-4.4026117E+00	S3	0.000000E+00	S3
2.1298373E+02	S2	-4.4037123E+00	S2	-1.1728784E+01	S2	-4.4026117E+00	S2
2.0149222E+02	S1	-1.9392389E-01	S1	-5.2570617E-01	S1	-1.1728784E+01	S1
3.7139160E+02	S0	-2.5919853E-03	S0	7.9218327E-05	S0	-5.2570617E-01	S0

NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
9.5547733E+00	S3	-4.8368412E-01	S3	-1.4663480E+01	S3	0.000000E+00	S3
6.8667969E+02	S2	-1.4944140E+01	S2	-5.2633209E+01	S2	-1.4663480E+01	S2
1.2178918E+03	S1	-6.1670989E-01	S1	-2.2647543E+00	S1	-5.2633209E+01	S1
1.6729531E+03	S0	2.2542518E-02	S0	4.3664919E-04	S0	-2.2647543E+00	S0

NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-4.0651712E+00	S3	-1.4863503E-01	S3	2.4954605E+01	S3	0.000000E+00	S3
-1.2204497E+03	S2	2.4573669E+01	S2	4.7512680E+01	S2	2.4954605E+01	S2
-6.3457568E+02	S1	1.1205330E+00	S1	2.2060394E+00	S1	4.7512680E+01	S1
-1.4958289E+03	S0	5.8258414E-02	S0	-2.0742866E-04	S0	2.2060394E+00	S0

NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN
-4.9463593E+01	S4	-4.6864380E+02	S4	-1.4401320E+02	S4
-1.0664063E+00	S3	-2.7193359E+02	S3	-3.6908984E+02	S3
1.1169188E+04	S2	5.0375746E+04	S2	-4.4909938E+04	S2
4.8821997E+02	S1	2.1325811E+03	S1	-2.0055359E+03	S1
-9.1167915E-01	S0	-4.0199451E+00	S0	3.7046003E+00	S0

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y	L	N
	FT/SEC**2	RAD/SEC**2	RAD/SEC**2
BETA (PER RAD)	-322.0231934	-60.0243330	13.2632236
P (PER R/S)	.0000000	-6.0789536	-.2134472
R (PER R/S)	.0000000	1.8890333	-.1063198
AILERON (PER RAD)	-91.9892883	129.1889101	8.4956694
RUDDER (PER RAD)	60.1864319	29.5793915	-8.1139545

A Matrix:

B Matrix:

BETA	P	R	PHI	AIL	RUD
-.3320E+00	.4936E-01	-.9988E+00	.3313E-01	-.9483E-01	.6204E-01
-.6002E+02	-.6079E+01	.1889E+01	.0000E+00	.1292E+03	.2958E+02
.1326E+02	-.2134E+00	-.1063E+00	.0000E+00	.8496E+01	-.8114E+01
.0000E+00	.1000E+01	.4942E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-2.8749512E-01 +J(3.9827080E+00)
 -2.8749512E-01 +J(-3.9827080E+00)
 -5.9503202E+00 +J(0.0000000E+00)
 8.1319362E-03 +J(0.0000000E+00)

DYNAMICS

PERIOD= 1.5776 WD= 3.98271
 ZETA= .07200 WN= 3.99307

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
6.5171261E+00 S3
1.9312714E+01 S2
9.4716585E+01 S1
-7.7154928E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-9.4825804E-02 S3	1.291892E+02 S3	8.4956694E+00 S3	0.000000E+00 S3
-2.6947126E+00 S2	7.8360397E+01 S2	2.5631378E+01 S2	1.2960880E+02 S2
-1.8375519E+01 S1	2.2288140E+03 S1	1.0888292E+02 S1	7.9627182E+01 S1
1.0260248E+00 S0	-3.6399517E+00 S0	7.3651794E-01 S0	2.2341953E+03 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
6.2042277E-02 S3	2.9579391E+01 S3	-8.1139545E+00 S3	0.000000E+00 S3
9.9479237E+00 S2	-6.0877457E+00 S2	-5.7507751E+01 S2	2.9178375E+01 S2
5.5999649E+01 S1	-9.7486969E+01 S1	-1.7347321E+01 S1	-8.9299489E+00 S1
-4.9467725E-01 S0	1.5523255E-01 S0	-3.1374454E+00 S0	-9.8344315E+01 S0

NUMERATOR AY/ AIL	NUMERATOR AY/ RUD
-9.1877182E+01 S4	6.0113052E+01 S4
-5.6579125E+02 S3	3.6049954E+02 S3
-8.8709180E+02 S2	-2.1073291E+03 S2
-2.7968125E+03 S1	-1.2331961E+04 S1
-2.5950928E+02 S0	1.1295226E+02 S0

AIRCRAFT STATE

MACH = 1.10000 ALT = 10000.00 VTREPS = 1185.66
 NZ = .99837 ALPHA = 2.43410 BETA = .00000
 GAMMA = .00000 THETA = 2.43411 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IYZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.78435 FLAPS = -3.60885 STRAKES= -4.59618
 AILERON= .00000 RUDDER = .00000
 THRUST = 13910.262 X RPM = 86.54823

DYNAMICS

CXAERO = -.05795356 CYAERO = .00000000 CZAERO = -.06968403
 CLMAERO= .00000000 CMMAERO= .00725945 CNMAERO= .00000000
 VC KTS = 625.694 DYN PR = 1234.01392 RHO = .00175561

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2562799	-7.3732319	.7643240	.0000000
PITCH RATE (PER RAD)	.0000000	-7.0114536	-9.1287146	.0000000
STRAKE (PER RAD)	.0044665	-.1124151	-.1950133	.0000000
MACH (PER M #)	-.0398733	.2354351	.0818803	3771.1708984
ALPHA DOT (PER RAD)	.0000000	-.4071600	-1.5345621	.0000000
CANARD (PER RAD)	-.0144666	-.7776472	.7617968	.0000000
FLAPERON (PER RAD)	.0278860	-.6557210	-.5147502	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1135759	.1853496	-.0750934
ROLL RATE (PER RAD)	.0000000	.0702794	-.6063483
YAW RATE (PER RAD)	.0000000	-.3829870	.2318162
AILERON (PER RAD)	-.3548137	.0773319	.0668529
RUDDER (PER RAD)	.0863965	-.0593754	.0158330

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-3392.9519043	25.5960388	114.8867187
ALPHADOT (PER R/S)	-.5708611	-.1555632	.0000000
Q (PER R/S)	-9.8482094	-.9270774	.0700000
VEL (PER FT/S)	-1683347	.0019655	-.0592737
STRAKE (PER RAD)	-51.8594666	-6.5046644	2.0605059
FLAPRN (PER RAD)	-302.4980469	-17.1694794	12.8643827
CANARD (PER RAD)	-358.7451172	25.4097137	-6.6737537

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.5927E-01	-.1149E+03	-.5036E+02	-.3214E+02	.2061E+01	.1286E+02	-.6674E+01
.1420E-03	-.2863E+01	.9912E+00	-.1153E-02	-.4374E-01	-.2552E+00	-.3027E+00
.1943E-02	.2604E+02	-.1081E+01	.1794E-03	-.6488E+01	-.1713E+02	-.2546E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-7.1260204E+00	,+J(0.0000000E+00)
3.1658268E+00	,+J(0.0000000E+00)
-1.3889593E-01	,+J(0.0000000E+00)
9.5684230E-02	,+J(0.0000000E+00)

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
 4.0034018E+00 S3
 -2.2401794E+01 S2
 -1.0270939E+00 S1
 2.9983733E-01 S0

NUMERATOR THT/STRK
 0.000000E+00 S3
 -6.4978571E+00 S2
 -2.0123077E+01 S1
 -1.0547905E+00 S0

NUMERATOR Q/STRK
 -6.4978571E+00 S3
 -2.0123077E+01 S2
 -1.0547905E+00 S1
 2.2875916E-05 S0

NUMERATOR ALP/STRK
 -4.3757215E-02 S3
 -6.4903488E+00 S2
 -3.3059347E-01 S1
 2.7413007E-02 S0

NUMERATOR U/STRK
 2.0605059E+00 S3
 3.3030029E+02 S2
 4.1078174E+02 S1
 6.3552344E+02 S0

NUMERATOR THT/FLAP
 0.000000E+00 S3
 -1.7129776E+01 S2
 -5.6677155E+01 S1
 -2.9589672E+00 S0

NUMERATOR Q/FLAP
 -1.7129776E+01 S3
 -5.6677155E+01 S2
 -2.9589672E+00 S1
 8.2715720E-05 S0

NUMERATOR ALP/FLAP
 -2.5523734E-01 S3
 -1.7268478E+01 S2
 -8.7868649E-01 S1
 6.3511431E-02 S0

NUMERATOR U/FLAP
 1.2864383E+01 S3
 8.8398804E+02 S2
 1.0801226E+03 S1
 1.7926955E+03 S0

NUMERATOR THT/CAN
 0.000000E+00 S3
 2.5456802E+01 S2
 6.6492432E+01 S1
 3.3076239E+00 S0

NUMERATOR Q/CAN
 2.5456802E+01 S3
 6.6492432E+01 S2
 3.3076239E+00 S1
 -1.3378855E-05 S0

NUMERATOR ALP/CAN
 -3.0269665E-01 S3
 2.4886810E+01 S2
 1.2213840E+00 S1
 -1.3701791E-01 S0

NUMERATOR U/CAN
 -6.6737537E+00 S3
 -1.3429795E+03 S2
 -1.0782729E+03 S1
 -2.0928640E+03 S0

NUMERATOR AN/STRK
 -5.1834488E+01 S4
 8.8945313E+00 S3
 2.3437160E+04 S2
 1.2544739E+03 S1
 -1.4683971E+00 S0

NUMERATOR AN/FLAP
 -3.0235229E+02 S4
 -1.6430469E+02 S3
 6.6075063E+04 S2
 3.5029617E+03 S1
 -4.1412086E+00 S0

NUMERATOR AN/CAN
 -3.5857227E+02 S4
 -6.7520703E+02 S3
 -7.7284875E+04 S2
 -3.9896414E+03 S1
 4.5354872E+00 S0

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y	L	N
	FT/SEC**2	RAD/SEC**2	RAD/SEC**2
BETA (PER RAD)	-513.7160645	-95.0792999	18.6086426
P (PER R/S)	.0000000	-9.5942173	-.2333662
R (PER R/S)	.0000000	3.4712467	-.3986934
AILERON (PER RAD)	-163.6831055	96.3195496	12.4935436
RUDDER (PER RAD)	39.8565521	19.0692596	-6.3572369

B Matrix:

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.4333E+00	-.4247E-01	-.9991E+00	.2711E-01	-.1381E+00	.3362E-01
-.9309E+02	-.9594E+01	.3471E+01	.0000E+00	.9632E+02	.1907E+02
-.1861E+02	-.2384E+00	-.3987E+00	.0000E+00	.1249E+02	-.6357E+01
.0000E+00	.1000E+01	.4251E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-9.3808517E+00 +J(0.0000000E+00)
 -5.2501249E-01 +J(4.6406145E+00)
 -5.2501249E-01 +J(-4.6406145E+00)
 4.6693087E-03 +J(0.0000000E+00)

DYNAMICS

PERIOD= 1.3540 WD= 4.64061
 ZETA= .11242 WN= 4.67022

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 1.0426184E+01 S3
 3.1612106E+01 S2
 2.0445616E+02 S1
 -9.5544773E-01 S0

NUMERATOR PHI/ AIL
 0.0000000E+00 S3
 9.6850616E+01 S2
 1.4086891E+02 S1
 3.0152925E+03 S0

NUMERATOR R/ AIL
 1.2493544E+01 S3
 9.9750595E+01 S2
 1.4079247E+02 S1
 8.0798737E+01 S0

NUMERATOR P/ AIL
 9.6319550E+01 S3
 1.3662868E+02 S2
 3.0093081E+03 S1
 -3.4341450E+00 S0

NUMERATOR BETA/ AIL
 -1.3805199E-01 S3
 -9.7711143E+00 S2
 -9.1362747E+01 S1
 2.3286266E+00 S0

NUMERATOR PHI/ RUD
 0.0000000E+00 S3
 1.8799011E+01 S2
 -1.2275104E+01 S1
 -2.5610376E+02 S0

NUMERATOR R/ RUD
 -6.3572369E+00 S3
 -6.7667068E+01 S2
 -3.2232559E+01 S1
 -6.7666521E+00 S0

NUMERATOR P/ RUD
 1.9069260E+01 S3
 -9.3986845E+00 S2
 -2.5473346E+02 S1
 2.8806537E-01 S0

NUMERATOR BETA/ RUD
 3.3615414E-02 S3
 7.4972879E+00 S2
 6.5530777E+01 S1
 -4.6772391E-01 S0

NUMERATOR /

NUMERATOR AY/ RUD
 3.9820587E+01 S4
 3.9026050E+02 S3
 -2.6616653E+03 S2
 -2.5514801E+04 S1
 2.0220926E+02 S0

NUMERATOR AY/ AIL
 -1.6353542E+02 S4
 -1.6252148E+03 S3
 -5.7023438E+01 S2
 1.3466000E+04 S1
 -1.0398855E+03 S0

AIRCRAFT STATE

MACH = .50000 ALT = 20000.00 VTRFPS = 516.69
 NZ = .99710 ALPHA = 5.39862 BETA = .00000
 GAMMA = .00000 THETA = 5.39865 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IYZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -3.29467 FLAPS = 6.93343 STRAKES= -6.84897
 AILERON= .00000 RUDDER = .00000
 THRUST = 1776.878 X RPM = 17.14639

DYNAMICS

CXAERO = -.00882808 CYAERO = .00000000 CZAERO = -.50259894
 CLMAERO= .00000000 CMMAERO= .00671974 CNMAERO= .00000000
 VC KTS = 227.978 DYN PR = 170.48216 RHO = .00126733

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.4467096	-4.8793049	1.6163874	.0000000
PITCH RATE(PER RAD)	.0000000	-6.7813272	-7.4650373	.0000000
STRAKE (PER RAD)	.0333331	-.2229946	-.3313628	.0000000
MACH (PER M #)	-.0055740	-.0170433	.0329146	670.4199219
ALPHA DOT (PER RAD)	.0000000	-.4774405	-1.0246820	.0000000
CANARD (PER RAD)	-.0408038	-.2019234	.8644326	.0000000
FLAPERON (PER RAD)	.0381018	-1.4610481	-.5628066	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0985174	.2724704	-.0451063
ROLL RATE (PER RAD)	.0000000	-.0244648	-.3524559
YAW RATE (PER RAD)	.0000000	-.2030303	.1798469
AILERON (PER RAD)	-.2164487	.0828704	.2274385
RUDDER (PER RAD)	.2489456	-.1458371	.0417726

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-314.2700195	7.3954010	28.0959320
ALPHADOT (PER R/S)	-.2099026	-.0325720	.0000000
Q (PER R/S)	-3.0079775	-.2394138	.0000000
VEL (PER FT/S)	-.0675997	-.0010979	-.0063593
STRAKE (PER RAD)	-14.2120428	-1.5269432	2.1244049
FLAPRN (PER RAD)	-93.1165009	-2.5934525	2.4283314
CANARD (PER RAD)	-12.8691177	3.9833670	-2.6005363

A Matrix:

B Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.6359E-02	.2810E+02	-.4880E+02	-.3203E+02	.2124E+01	.2428E+01	-.2601E+01
-.1309E-03	-.6083E+00	.9938E+00	-.5860E-02	-.2751E-01	-.1802E+00	-.2491E-01
-.1094E-02	.7415E+01	-.2718E+00	.1909E-03	-.1526E+01	-.2588E+01	-.3984E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

2.2938814E+00 /+J(0.0000000E+00)
-3.1676865E+00 /+J(0.0000000E+00)
-6.3389465E-03 /+J(8.4609926E-02)
-6.3389465E-03 /+J(-8.4609926E-02)

DYNAMICS

PERIOD= 74.2606 WD= .08461
ZETA= .07471 WN= .08485

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
8.8648432E-01 S3
-7.2479744E+00 S2
-8.5792065E-02 S1
-5.2293658E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
2.1244049E+00	S3	-2.7510688E-02	S3	-1.5260468E+00	S3	0.000000E+00	S3
7.5568649E+01	S2	-1.5244713E+00	S2	-1.1443844E+00	S2	-1.5260468E+00	S2
4.6018127E+01	S1	-1.1405565E-02	S1	-1.5440047E-02	S1	-1.1443844E+00	S1
3.6614212E+01	S0	-5.3612925E-03	S0	2.0166237E-07	S0	-1.5440047E-02	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
2.4283314E+00	S3	-1.8024844E-01	S3	-2.5875816E+00	S3	0.000000E+00	S3
1.2334828E+02	S2	-2.6219158E+00	S2	-2.9298267E+00	S2	-2.5875816E+00	S2
1.3381052E+02	S1	-1.1097185E-02	S1	-2.6455391E-02	S1	-2.9298267E+00	S1
9.3766068E+01	S0	-4.4180788E-03	S0	8.6339224E-07	S0	-2.6455391E-02	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-2.6003363E+00	S3	-2.4911147E-02	S3	3.9841785E+00	S3	0.000000E+00	S3
-1.9741864E+02	S2	3.9527712E+00	S2	2.2672024E+00	S2	3.9841785E+00	S2
-1.0709811E+02	S1	3.1485144E-02	S1	3.3903100E-02	S1	2.2672024E+00	S1
-7.2487122E+01	S0	1.7404839E-02	S0	-1.4674549E-07	S0	3.3903100E-02	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-1.4206268E+01	S4	-9.3078644E+01	S4	-1.2863889E+01	S4		
8.1347656E-01	S3	-1.7729736E+01	S3	-1.6218262E+01	S3		
5.8044067E+02	S2	1.4993726E+03	S2	-1.1424438E+03	S2		
1.7404633E+00	S1	2.5111246E+00	S1	-1.6566048E+00	S1		
-4.6841983E-02	S0	-8.0527723E-02	S0	1.0270226E-01	S0		

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-70.0114746	-6.8176937	4.2136688
P (PER R/S)	.0000000	-1.7725754	-.0721275
R (PER R/S)	.0000000	.8668829	-.0568079
AILERON (PER RAD)	-13.7948589	44.0713043	2.8864584
RUDDER (PER RAD)	15.8659735	7.0225477	-2.1379986

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.1350E+00	.9408E-01	-.9956E+00	.6175E-01	-.2660E-01	.3059E-01
-.6818E+01	-.1773E+01	.8669E+00	.0000E+00	.4407E+02	.7023E+01
.4214E+01	-.7213E-01	-.5681E-01	.0000E+00	.2886E+01	-.2138E+01
.0000E+00	.1000E+01	.9450E-01	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-1.4543992E-01 ,+J(2.1886539E+00)
-1.4543992E-01 ,+J(-2.1886539E+00)
-1.7037611E+00 ,+J(0.0000000E+00)
3.0267972E-02 ,+J(0.0000000E+00)

DYNAMICS

PERIOD= 2.8708 WD= 2.18865
ZETA= .06631 WN= 2.19348

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
1.964360E+00 S3
5.246559E+00 S2
8.036707E+00 S1
-2.481155E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-2.659548E-02 S3	4.407130E+01 S3	2.836453E+00 S3	0.000000E+00 S3
1.224089E+00 S2	1.113575E+01 S2	2.215253E+00 S2	4.434407E+01 S2
1.275930E+00 S1	2.050586E+02 S1	1.937283E+01 S1	1.134509E+01 S1
3.204371E-01 S0	-1.198624E+00 S0	1.268308E+01 S0	2.068894E+02 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
3.058844E-02 S3	7.022547E+00 S3	-2.137998E+00 S3	0.000000E+00 S3
2.845180E+00 S2	-7.151188E-01 S2	-4.455973E+00 S2	6.820499E+00 S2
4.566563E+00 S1	1.485143E+01 S1	1.076228E+00 S1	-1.136221E+00 S1
-1.148930E-01 S0	-8.762508E-02 S0	9.272042E-01 S0	1.495314E+01 S0

AIRCRAFT STATE

MACH =	.70000	ALT =	20000.00	VTRFPS =	726.17
NZ =	.99946	ALPHA =	3.65152	BETA =	.00000
GAMMA =	.00000	THETA =	3.65153	ACGW =	15926.00
IXX =	4548.00	IYY =	49429.00	IZZ =	52531.00
IXZ =	1827.00	XCG =	450.56006	ZCG =	64.88000
GEAR DOWN:	F	NY =	.00000	NX =	.00000
NORMAL MODE:	T	DR MODE:	F	AR MODE:	F

CONTROLS

CANARDS=	-2.81643	FLAPS =	1.56063	STRAKES=	-3.98842
AILERON=	.00000	RUDDER =	.00000		
THRUST =	2178.077	X RPM =	19.26140		

DYNAMICS

CXAERO =	-.01882118	CYAERO =	.000000000	CZAERO =	-.25705642
CLMAERO=	.000000000	CMMAERO=	.00420278	CNMAERO=	.000000000
VC KTS =	323.811	DYN PR =	334.14453	RHO =	.00126733

A/C CONSTANTS

AREA =	185.0490	CBAR =	7.2200	SPAN =	27.2000
--------	----------	--------	--------	--------	---------

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.3071296	-5.1370735	1.6856546	.0000000
PITCH RATE(PER RAD)	.0000000	-7.2728968	-7.3926697	.0000000
STRAKE (PER RAD)	.0226788	-2.110947	-2.2995498	.0000000
MACH (PER 1 #)	.0081028	-0.0889266	.0235080	1067.0725098
ALPHA DOT (PER RAD)	.0000000	-2.4740056	-1.1184645	.0000000
CANARD (PER RAD)	-0.0063362	-2.2993854	.8918194	.0000000
FLAPERON (PER RAD)	.0358100	-1.5469923	-2.5789520	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
SETA (PER RAD)	-1.0752907	.2294621	-0.0507144
ROLL RATE (PER RAD)	.0000000	-0.0145733	-0.3756201
YAW RATE (PER RAD)	.0000000	-0.1834183	.1569413
AILERON (PER RAD)	-0.2179332	.0768455	.2052163
RUDDER (PER RAD)	.2449960	-0.1434140	.0419127

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-643.6755371	15.1770391	37.9559479
ALPHADOT (PER R/S)	-.2931610	-.0500152	.0000000
Q (PER R/S)	-4.5164280	-.3543798	.0000000
VEL (PER FT/S)	-.0426657	-.0010446	-.0067752
STRAKE (PER RAD)	-26.3690796	-2.7054768	2.8329391
FLAPRN (PER RAD)	-193.2439880	-5.2289829	4.4732342
CANARD (PER RAD)	-37.3979950	8.0547419	-.7914922

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.6775E-02	-.3796E+02	-.4625E+02	-.3211E+02	.2833E+01	.4473E+01	-.7915E+00
-.5885E-04	-.8878E+00	.9934E+00	-.2826E-02	-.3637E-01	-.2665E+00	-.5158E-01
-.1042E-02	-.1522E+02	-.4041E+00	.1414E-03	-.2704E+01	-.5216E+01	.8057E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-4.5473042E+00 /+J(0.0000000E+00)
3.2580271E+00 /+J(0.0000000E+00)
-4.7118999E-03 /+J(6.2520385E-02)
-4.7118999E-03 /+J(-6.2520385E-02)

DYNAMICS

PERIOD=100.4982 WD= .06252
ZETA= .07515 WN= .06270

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
1.2985822E+00 S3
-1.4799049E+01 S2
-1.3458127E-01 S1
-5.8279234E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
2.8329391E+00 S3		-3.6371797E-02 S3		-2.7036572E+00 S3		0.0000000E+00 S3	
1.2731752E+02 S2		-2.7008314E+00 S2		-2.9753256E+00 S2		-2.7036572E+00 S2	
7.9113052E+01 S1		-1.9253116E-02 S1		-2.9772215E-02 S1		-2.9753256E+00 S1	
9.5262604E+01 S0		-3.8324371E-03 S0		5.2135792E-06 S0		-2.9772215E-02 S0	
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
4.4732342E+00 S3		-2.6654822E-01 S3		-5.2156515E+00 S3		0.0000000E+00 S3	
2.3687425E+02 S2		-5.2908230E+00 S2		-8.7279301E+00 S2		-5.2156515E+00 S2	
3.0249219E+02 S1		-2.7141154E-02 S1		-6.8115532E-02 S1		-8.7279301E+00 S1	
2.7971069E+02 S0		-8.2753203E-04 S0		9.7316015E-06 S0		-6.8115532E-02 S0	
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-7.9149216E-01 S3		-5.1584367E-02 S3		8.0573215E+00 S3		0.0000000E+00 S3	
-3.7561377E+02 S2		7.9827232E+00 S2		6.4238691E+00 S2		8.0573215E+00 S2	
-2.3854962E+02 S1		5.6569908E-02 S1		6.4623952E-02 S1		6.4238691E+00 S1	
-2.0538074E+02 S0		1.6794771E-02 S0		-1.5797719E-05 S0		6.4623952E-02 S0	
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-2.6358414E+01 S4		-1.9316585E+02 S4		-3.7382874E+01 S4			
2.0478516E+00 S3		-5.4476318E+01 S3		-5.4058594E+01 S3			
2.1367075E+03 S2		6.2947227E+03 S2		-4.5978320E+03 S2			
1.2701723E+01 S1		3.0878983E+01 S1		-2.1498489E+01 S1			
-6.4783871E-02 S0		-1.4662659E-01 S0		1.4386815E-01 S0			

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y	L	N
	FT/SEC**2	RAD/SEC**2	RAD/SEC**2
BETA (PER RAD)	-134.320823	-16.0269318	6.7891464
P (PER R/S)	.0000000	-2.6418896	-.1006219
R (PER R/S)	.0000000	1.0575371	-.0732003
AILERON (PER RAD)	-27.2233276	77.9667969	5.1719599
RUDDER (PER RAD)	30.6038971	13.8483038	-4.1099663

B Matrix:

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.1850E+00	.6369E-01	-.9980E+00	.4422E-01	-.3749E-01	.4214E-01
-.1603E+02	-.2642E+01	.1058E+01	.0000E+00	.7797E+02	.1385E+02
.6789E+01	-.1006E+00	-.7320E-01	.0000E+00	.5172E+01	-.4110E+01
.0000E+00	.1000E+01	.6382E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.7012846E-01 ,+J(2.7819939E+00)
 -1.7012846E-01 ,+J(-2.7819939E+00)
 -2.5758400E+00 ,+J(0.0000000E+00)
 1.6029797E-02 ,+J(0.0000000E+00)

DYNAMICS

PERIOD= 2.2585 WD= 2.78199
 ZETA= .06104 WN= 2.78719

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 2.9000616E+00 S3
 8.5980930E+00 S2
 1.9871552E+01 S1
 -3.2076275E-01 S0

NUMERATOR PHI/ AIL
 0.0000000E+00 S3
 7.8296844E+01 S2
 2.6615341E+01 S1
 6.1532813E+02 S0

NUMERATOR R/ AIL
 5.1719599E+00 S3
 6.5207291E+00 S2
 3.9334076E+01 S1
 2.7070175E+01 S0

NUMERATOR P/ AIL
 7.7966797E+01 S3
 2.6199219E+01 S2
 6.1281787E+02 S1
 -1.7281036E+00 S0

NUMERATOR BETA/ AIL
 -3.7489023E-02 S3
 -2.9773617E-01 S2
 -1.6441708E+00 S1
 5.1061648E-01 S0

NUMERATOR PHI/ RUD
 0.0000000E+00 S3
 1.3586018E+01 S2
 -2.2587538E+00 S1
 2.7749908E+01 S0

NUMERATOR R/ RUD
 -4.1099663E+00 S3
 -1.2725623E+01 S2
 3.5036480E-01 S1
 1.2446213E+00 S0

NUMERATOR P/ RUD
 1.3848304E+01 S3
 -1.4466429E+00 S2
 2.7727554E+01 S1
 -7.9489350E-02 S0

NUMERATOR BETA/ RUD
 4.2144377E-02 S3
 5.0980120E+00 S2
 1.2627754E+01 S1
 -1.8193901E-01 S0

NUMERATOR /

NUMERATOR AY/ RUD
 3.0541763E+01 S4
 7.5577148E+01 S3
 -4.4025220E+02 S2
 -1.0877554E+03 S1
 1.4632750E+01 S0

NUMERATOR AY/ AIL
 -2.7168060E+01 S4
 -7.3471924E+01 S3
 -1.9165356E+02 S2
 -3.2084766E+02 S1
 -5.9860275E+01 S0

AIRCRAFT STATE

MACH = .90000 ALT = 20000.00 VTRFPS = 933.64
 N2 = .99733 ALPHA = 3.23396 BETA = .00000
 GAMMA = .00000 THETA = 3.23398 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 Ixz = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -3.58662 FLAPS = -2.28628 STRAKES= -2.30955
 AILERON= .00000 RUDDER = .00000
 THRUST = 3457.378 X RPM = 28.48160

DYNAMICS

CXAERO = -.02503515 CYAERO = .00000000 CZAERC = -.15556610
 CLMAERO= .00000000 CMMAERO= .00403757 CNMAERO= .00000000
 VC KTS = 423.744 DYN PR = 552.36182 RHO = .00126733

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.3094575	-6.1562443	1.7845736	.0000000
PITCH RATE(PER RAD)	.0000000	-8.0021572	-8.8420544	.0000000
STRAKE (PER RAD)	.0062426	-.1809973	-.2213879	.0000000
MACH (PER M #)	-.0742142	-.2927207	-.0782672	788.9379883
ALPHA DOT (PER RAD)	.0000000	-.4486861	-1.2793741	.0000000
CANARD (PER RAD)	-.0091512	-.4128456	1.0362921	.0000000
FLAPERON (PER RAD)	.0309399	-1.5183430	-.6590123	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0620575	.2079553	-.0725927
ROLL RATE (PER RAD)	.0000000	-.0060346	-.4262069
YAW RATE (PER RAD)	.0000000	-.1580547	.1641418
AILERON (PER RAD)	-.2978523	.0513388	.1499838
RUDDER (PER RAD)	.2176202	-.1295782	.0405709

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-1273.8625488	26.5066986	62.3384957
ALPHADOT (PER R/S)	-.3571006	-.0736212	.0000000
Q (PER R/S)	-6.3890972	-.5104374	.0000000
VEL (PER FT/S)	-.0500675	-.0026188	-.0281302
STRAKE (PER RAD)	-37.3748169	-3.3053532	1.2890587
FLAPRN (PER RAD)	-313.5283203	-9.8391466	6.388988
CANARD (PER RAD)	-85.2500305	15.4719772	-1.8896675

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.2813E-01	.6234E+02	-.5267E+02	-.3212E+02	.1289E+01	.6389E+01	-.1890E+01
-.5369E-04	-.1366E+01	.9923E+00	-.1946E-02	-.4008E-01	-.3362E+00	-.9142E-01
-.2615E-02	.2661E+02	-.5835E+00	.1433E-03	-.3302E+01	-.9814E+01	.1548E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-6.1406317E+00 +J(0.0000000E+00)
 4.1964583E+00 +J(0.0000000E+00)
 -1.6768601E-02 +J(7.6866567E-02)
 -1.6768601E-02 +J(-7.6866567E-02)

DYNAMICS

PERIOD= 81.7415 WD= .07687
 ZETA= .21314 WN= .07867

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 1.9777060E+00 S3
 -2.5697342E+01 S2
 -8.5262746E-01 S1
 -1.5944296E-01 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
1.2890587E+00	S3	-4.0079590E-02	S3	-3.3024025E+00	S3	0.0000000E+00	S3
1.7395123E+02	S2	-3.3030958E+00	S2	-5.6739235E+00	S2	-3.3024025E+00	S2
1.6099756E+02	S1	-9.3652129E-02	S1	-1.6784680E-01	S1	-5.6739235E+00	S1
1.7963637E+02	S0	-2.1264625E-03	S0	3.0512776E-05	S0	-1.6784680E-01	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
6.3888988E+00	S3	-3.3621800E-01	S3	-9.8143940E+00	S3	0.0000000E+00	S3
5.0841772E+02	S2	-9.9493856E+00	S2	-2.2645584E+01	S2	-9.8143940E+00	S2
7.0929248E+02	S1	-2.5867540E-01	S1	-6.3871610E-01	S1	-2.2645584E+01	S1
7.1955396E+02	S0	1.1931516E-02	S0	1.1690304E-04	S0	-6.3871610E-01	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-1.8896675E+00	S3	-9.1419458E-02	S3	1.5478707E+01	S3	0.0000000E+00	S3
-3.2454233E+02	S2	1.5310912E+01	S2	1.9152618E+01	S2	1.5478707E+01	S2
-4.7976636E+02	S1	4.6196872E-01	S1	6.0245264E-01	S1	1.9152618E+01	S1
-6.0306079E+02	S0	3.3448938E-02	S0	-1.0927141E-04	S0	6.0245264E-01	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-3.7360504E+01	S4	-3.1340820E+02	S4	-8.5217377E+01	S4		
-6.4624023E-01	S3	-1.2583203E+02	S3	-1.5641406E+02	S3		
5.1957031E+03	S2	2.0850324E+04	S2	-1.7394543E+04	S2		
1.4417918E+02	S1	5.6540381E+02	S1	-4.9563367E+02	S1		
-3.3309013E-01	S0	-1.2682629E+00	S0	1.1953287E+00	S0		

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-219.3083496	-40.5210724	9.5967598
P (PER R/S)	.0000000	-3.8508730	-.1358336
R (PER R/S)	.0000000	1.4326839	-.0720223
AILERON (PER RAD)	-61.5046539	94.0917511	5.9895716
RUDDER (PER RAD)	44.9372101	22.3585815	-6.0803213

3 Matrix:

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.2349E+00	-.5641E-01	-.9984E+00	.3441E-01	-.6588E-01	-.4813E-01
-.4052E+02	-.3951E+01	.1433E+01	.0000E+00	.9409E+02	.2236E+02
.9597E+01	-.1386E+00	-.7202E-01	.0000E+00	.5990E+01	-.6080E+01
.0000E+00	.1000E+01	.5650E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-2.1961498E-01 ,J(3.4091587E+00)
 -2.1961498E-01 ,J(-3.4091587E+00)
 -3.7289696E+00 ,J(0.0000000E+00)
 1.0464229E-02 ,J(0.0000000E+00)

DYNAMICS

PERIOD= 1.8430 WD= 3.40916
 ZETA= .06429 WN= 3.41623

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 4.1577892E+00 S3
 1.3264754E+01 S2
 4.3379944E+01 S1
 -4.5542973E-01 S0

NUMERATOR PHI/ AIL
 0.0000000E+00 S3
 9.4430170E+01 S2
 4.0739044E+01 S1
 1.1503757E+03 S0

NUMERATOR R/ AIL
 5.9595716E+00 S3
 1.0800224E+01 S2
 6.4181763E+01 S1
 3.9417877E+01 S0

NUMERATOR P/ AIL
 9.4091751E+01 S3
 4.0128799E+01 S2
 1.1467493E+03 S1
 -2.2291775E+03 S0

NUMERATOR BETA/ AIL
 -6.5375888E-02 S3
 -9.3045735E-01 S2
 -5.9235648E+00 S1
 5.4789352E-01 S0

NUMERATOR PHI/ RUD
 0.0000000E+00 S3
 2.2015015E+01 S2
 -5.3519545E+00 S1
 -3.3244690E+01 S0

NUMERATOR R/ RUD
 -6.0803213E+00 S3
 -2.7479416E+01 S2
 -5.9733315E+00 S1
 -1.0944576E+00 S0

NUMERATOR P/ RUD
 2.2358592E+01 S3
 -3.7992849E+00 S2
 -3.2907181E+01 S1
 6.1635610E-02 S0

NUMERATOR BETA/ RUD
 4.8130952E-02 S3
 7.5207663E+00 S2
 2.6850616E+01 S1
 -2.9586720E-01 S0

NUMERATOR /

NUMERATOR AY/ RUD
 4.4865646E+01 S4
 1.6510229E+02 S3
 -1.0932170E+03 S2
 -3.9387527E+03 S1
 4.4457291E+01 S0

NUMERATOR AY/ AIL
 -6.1406738E+01 S4
 -2.3989063E+02 S3
 -6.0297900E+02 S2
 -1.3693555E+03 S1
 -9.2144531E+01 S0

AIRCRAFT STATE

WACH = 1.10000 ALT = 20000.00 VTRFPS = 1141.12
 NZ = .99796 ALPHA = 2.69010 BETA = .00000
 GAMMA = .00000 THETA = 2.69012 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IYZ = 1827.00 XCG = 450.56006 ZCG = 64.89000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.74969 FLAPS = -2.59503 STRAKES= -5.43995
 AILERON= .00000 RUDDER = .00000
 THRUST = 9265.582 % RPM = 70.44980

DYNAMICS

CXAERO = -.05578771 CYAERO = .00000000 CZAERO = -.10421002
 CLMAERO= .00000000 CMMAERO= .00726094 CNMAERO= .00000000
 VC KTS = 527.733 DYN PR = 825.13257 RHO = .00126733

A/C CONSTANTS

AREA = 185.0490 C9AR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2548127	-6.6815357	.8093599	.0000000
PITCH RATE (PER RAD)	.0000000	-7.0114584	-9.1286259	.0000000
STRAKE (PER RAD)	.0065848	-.1366521	-.2426434	.0000000
MACH (PER M #)	-.0420622	.1669449	.0470548	3233.0952051
ALPHA DOT (PER RAD)	.0000000	-.4069776	-1.5346003	.0000000
CANARD (PER RAD)	-.0179036	-.6994841	.8232118	.0000000
FLAPERON (PER RAD)	.0279860	-.7601296	-.5701944	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1401815	.2041374	-.0795999
ROLL RATE (PER RAD)	.0000000	.0676604	-.5329912
YAW RATE (PER RAD)	.0000000	-.3941029	.2338768
AILERON (PER RAD)	-.3550310	.0766411	.0886888
RUDDER (PER RAD)	.1091133	-.0721265	.0217854

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-2056.8457031	18.0807495	76.1452789
ALPHADOT (PER R/S)	-.3962740	-.1080381	.0000000
Q (PER R/S)	-6.8421249	-.6440882	.0000000
VEL (PER FT/S)	-.0780774	.0004974	-.0395579
STRAKE (PER RAD)	-42.1525421	-5.4116945	2.0311737
FLAPRN (PER RAD)	-234.4741821	-12.7170849	8.6018715
CANARD (PER RAD)	-215.7671204	18.3601532	-5.5226517

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.3956E-01	.7615E+02	-.5356E+02	-.3214E+02	.2031E+01	-.8602E+01	-.5523E+01
.6847E-04	-.1804E+01	.9937E+00	-.1324E-02	-.3697E-01	-.2056E+00	-.1892E+00
.4900E-03	.1828E+02	-.7514E+00	.1431E-03	-.5408E+01	-.1269E+02	.1838E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

3.0085392E+00	+J(0.0000000E+00)
-5.5711107E+00	+J(0.0000000E+00)
-8.2584739E-02	+J(0.0000000E+00)
5.0300807E-02	+J(0.0000000E+00)

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
2.5948362E+00 S3
-1.6682190E+01 S2
-5.5167902E-01 S1
6.9664717E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
2.0311737E+00	S3	-3.6967482E-02	S3	-5.4077005E+00	S3	0.0000000E+00	S3
2.9199512E+02	S2	-5.4024763E+00	S2	-1.0643142E+01	S2	-5.4077005E+00	S2
2.8700146E+02	S1	-1.8653184E-01	S1	-3.8140793E-01	S1	-1.0643142E+01	S1
3.3580469E+02	S0	1.1627290E-02	S0	3.3793433E-05	S0	-3.8140793E-01	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
8.6018715E+00	S3	-2.0563221E-01	S3	-1.2694859E+01	S3	0.0000000E+00	S3
6.8622070E+02	S2	-1.2776352E+01	S2	-2.7155518E+01	S2	-1.2694859E+01	S2
7.1835864E+02	S1	-4.4246387E-01	S1	-9.7754383E-01	S1	-2.7155518E+01	S1
8.5821899E+02	S0	2.5422353E-02	S0	9.0195914E-05	S0	-9.7754383E-01	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-5.5226517E+00	S3	-1.8922627E-01	S3	1.8380600E+01	S3	0.0000000E+00	S3
-1.0129297E+03	S2	1.8113861E+01	S2	3.0421768E+01	S2	1.8380600E+01	S2
-7.0853906E+02	S1	6.1718333E-01	S1	1.0599604E+00	S1	3.0421768E+01	S1
-9.5640913E+02	S0	-4.4478212E-02	S0	-8.6041662E-05	S0	1.0599604E+00	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-4.2137909E+01	S4	-2.3439276E+02	S4	-2.1569221E+02	S4		
5.9531250E+00	S3	-9.2878906E+01	S3	-3.0404297E+02	S3		
1.1910949E+04	S2	3.0430082E+04	S2	-3.3945422E+04	S2		
4.3193506E+02	S1	1.1022393E+03	S1	-1.2129717E+03	S1		
-6.1446208E-01	S0	-1.5789433E+00	S0	1.6986609E+00	S0		

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y	L	N
	FT/SEC**2	RAD/SEC**2	RAD/SEC**2
BETA (PER RAD)	-351.7072754	-67.1439056	13.8040886
P (PER R/S)	.0000000	-5.8569746	-.1399490
R (PER R/S)	.0000000	2.4301519	-.2863269
AILERON (PER RAD)	-109.5149994	84.6052399	9.0018473
RUDDER (PER RAD)	33.6577301	17.9527527	-5.0814829

A Matrix:

S Matrix:

BETA	P	Q	PHI	AIL	RUD
-.3082E+00	.4693E-01	-.9989E+00	.2816E-01	-.9597E-01	.2950E-01
-.6714E+02	-.5857E+01	.2430E-01	.0000E+00	.8461E+02	-.785E+02
.1380E+02	-.1399E+00	-.2868E+00	.0000E+00	.9002E+01	-.5081E+01
.0000E+00	.1000E+01	.4699E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-4.0558958E-01	+JC	4.0180931E+00)
-4.0558958E-01	+JC	-4.0180931E+00)
-5.6464739E+00	+JC	0.0000000E+00)
5.6662001E-03	+JC	0.0000000E+00)

DYNAMICS

PERIOD=	1.5637	WD=	4.01809
ZETA=	.10043	WN=	4.03351

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
6.4520130E+00 S3
2.085380E+01 S2
9.1972260E+01 S1
-5.2186096E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-9.5971406E-02 S3	3.4605240E+01 S3	9.0018473E+00 S3	0.000000E+00 S3
-5.6107264E+00 S2	7.8663116E+01 S2	4.2332840E+01 S2	8.5028193E+01 S2
-3.6471603E+01 S1	1.7832146E+03 S1	8.7120773E+01 S1	8.0652130E+01 S1
1.3536968E+00 S0	-2.3459644E+00 S0	4.9915405E+01 S0	1.7873079E+03 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ FUD
2.9495306E-02 S3	1.7852753E+01 S3	-5.0814829E+00 S3	0.000000E+00 S3
6.0949898E+00 S2	-3.7061319E+00 S2	-3.3419617E+01 S2	1.7613983E+01 S2
3.2441452E+01 S1	-9.6451645E+01 S1	-1.1728155E+01 S1	-5.2763662E+00 S1
-2.4629128E-01 S0	1.2548792E-01 S0	-2.6684589E+00 S0	-9.7002701E+01 S0

NUMERATOR AY/ AIL	NUMERATOR AY/ RUD
-1.0939435E+02 S4	3.3620636E+01 S4
-6.6578125E+02 S3	1.9912256E+02 S3
-2.6462891E+02 S2	-1.4825732E+03 S2
2.7532500E+03 S1	-8.3140117E+03 S1
-4.1905249E+02 S0	6.9124222E+01 S0

AIRCRAFT STATE

MACH = 1.20000 ALT = 20000.00 VTRFPS = 1244.86
 NZ = .99853 ALPHA = 2.58764 BETA = .00000
 GAMMA = .00000 THETA = 2.58765 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IYZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .03000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.26038 FLAPS = -1.09730 STRAKES= -7.00000
 AILERON= .00000 RUDDER = .00000
 THRUST = 11813.973 % RPM = 87.10735

DYNAMICS

CXAERO = -.06108531 CYAERO = .00000000 CZAERO = -.08756304
 CLMAERO= .00000000 CMMAERO= .00776303 CNMAERO= .00000000
 VC KTS = 579.107 DYN PR = 981.97510 RHO = .00126733

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2187977	-6.3249683	.5628051	.0000000
PITCH RATE(PER RAD)	.0000000	-6.1532333	-8.5496150	.0000000
STRAKE (PER RAD)	.0110895	-.1250732	-.2317507	.0000000
MACH (PER M #)	-.0431444	.1197350	-.0851833	4152.0312500
ALPHA DOT (PER RAD)	.0000000	-.3862703	-1.5186577	.0000000
CANARD (PER RAD)	-.0163906	-.5097920	.6997954	.0000000
FLAPERON (PER RAD)	.0263562	-.5748703	-.5274718	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1646433	.2105690	-.0883488
ROLL RATE (PER RAD)	.0000000	.0704445	-.5611741
YAW RATE (PER RAD)	.0000000	-.4563156	.2216032
AILERON (PER RAD)	-.3384541	.0977540	.0604245
RUDDER (PER RAD)	.0729126	-.0509129	.0144149

LONGITUDINAL SYSTEM
 BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	FT/SEC**2	M	RAD/SEC**2	X	FT/SEC**2
ALPHA (PER RAD)	-2317.5823730		14.8040037		77.2769775	
ALPHADOT (PER R/S)	-4103707		-1166546		.0000000	
Q (PER R/S)	-6.5505037		-6580738		.0000000	
VEL (PER FT/S)	-0749447		-0024580		-3460699	
STRAKE (PER RAD)	-45.9143219		-6.1512319		4.0709496	
FLAPRN (PER RAD)	-211.0346527		-14.0003929		9.6753521	
CANARD (PER RAD)	-187.1444397		18.5742798		-6.0169321	

B Matrix:

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-4607E-01	.7728E+02	-.5620E+02	-.3214E+02	.4071E+01	.9675E+01	-.6017E+01
.6024E-04	-.1863E+01	.9944E+00	-.1168E-02	-.3691E-01	-.1696E+00	-.1504E+00
-.2465E-02	.1502E+02	-.7741E+00	.1362E-03	-.6147E+01	-.1398E+02	.1859E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-5.2386503E+00	+/J(0.0000000E+00)
2.6086397E+00	+/J(0.0000000E+00)
-2.6627243E-02	+/J(8.9001060E-02)
-2.6627243E-02	+/J(-8.9001060E-02)

DYNAMICS

PERIOD= 70.5968	WD= .03900
ZETA= .28663	WN= .09290

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 2.6832342E+00 S3
 -1.3516954E+01 S2
 -7.0509624E-01 S1
 -1.1794496E-01 S0

NUMERATOR THT/STRK
 0.0000000E+00 S3
 -6.1469259E+00 S2
 -1.2299903E+01 S1
 -5.3250802E-01 S0

NUMERATOR Q/STRK
 -6.1469259E+00 S3
 -1.2299903E+01 S2
 -5.3250802E-01 S1
 1.2370465E-06 S0

NUMERATOR ALP/STRK
 -3.6903619E-02 S3
 -6.1425571E+00 S2
 -2.5960004E-01 S1
 1.5171483E-02 S0

NUMERATOR THT/FLAP
 0.0000000E+00 S3
 -1.3980603E+01 S2
 -2.9263290E+01 S1
 -1.2556553E+00 S0

NUMERATOR Q/FLAP
 -1.3980603E+01 S3
 -2.9263290E+01 S2
 -1.2556553E+00 S1
 3.2566085E-06 S0

NUMERATOR ALP/FLAP
 -1.6964197E-01 S3
 -1.4040924E+01 S2
 -5.8260912E-01 S1
 4.1297406E-02 S0

NUMERATOR THT/CAN
 0.0000000E+00 S3
 1.5591827E+01 S2
 3.3249786E+01 S1
 1.4559526E+00 S0

NUMERATOR Q/CAN
 1.8591827E+01 S3
 3.3249786E+01 S2
 1.4559526E+00 S1
 -2.3512675E-06 S0

NUMERATOR ALP/CAN
 -1.5043765E-01 S3
 1.8364044E+01 S2
 7.9703683E-01 S1
 -2.5104310E-02 S0

NUMERATOR AN/CAN
 -1.8708273E+02 S4
 -2.3326953E+02 S3
 -4.0330906E+04 S2
 -1.7935298E+03 S1
 2.1177912E+00 S0

NUMERATOR AN/FLAP
 -2.1096503E+02 S4
 -7.5015625E+01 S3
 3.5646695E+04 S2
 1.5703704E+03 S1
 -1.8279848E+00 S0

NUMERATOR AN/STRK
 -4.5399185E+01 S4
 5.4335938E+00 S3
 1.4964273E+04 S2
 6.6322241E+02 S1
 -7.7504158E-01 S0

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-427.5400391	-89.3029022	16.7063446
P (PER R/S)	.0000000	-6.7276125	-1615717
R (PER R/S)	.0000000	2.4772358	-3828979
AILERON (PER RAD)	-124.2463684	70.3446198	11.6441364
RUDDER (PER RAD)	26.7661896	13.9359350	-4.3056631

A Matrix:

	P	R	PHI	AIL	RUD
BETA					
-3434E+00	.4515E-01	-.9990E+00	.2582E-01	-.9981E-01	.2150E-01
-.8930E+02	-.6728E+01	.2477E+01	.0000E+00	.7034E+02	.1394E+02
.1671E+02	-.1616E+00	-.3829E+00	.0000E+00	.1164E+02	-.4306E+01
.0000E+03	.1000E+01	.4519E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-4.7717381E-01 ,+J(4.4414616E+00)
 -4.7717381E-01 ,+J(-4.4414616E+00)
 -6.5022373E+00 ,+J(0.0000000E+00)
 2.5716668E-03 ,+J(0.0000000E+00)

DYNAMICS

PERIOD= 1.4147 WD= 4.44146
 ZETA= .10682 WN= 4.46702

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
7.4539547E+00 S3
2.6139389E+01 S2
1.2967703E+02 S1
-3.3369493E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-9.9807620E-02 S3	7.0344620E+01 S3	1.1644136E+01 S3	0.0000000E+00 S3
-9.1660843E+00 S2	8.8852646E+01 S2	6.9303223E+01 S2	7.0870850E+01 S2
-6.2352173E+01 S1	2.2312373E+03 S1	1.1034673E+02 S1	9.1084680E+01 S1
1.5183773E+00 S0	-2.5842113E+00 S0	5.7190796E+01 S0	2.2362244E+03 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
2.1501381E-02 S3	1.3935935E+01 S3	-4.3056631E+00 S3	0.0000000E+00 S3
5.0833273E+00 S2	-2.4640217E+00 S2	-3.2338043E+01 S2	1.3741347E+01 S2
3.1364792E+01 S1	-1.5321103E+02 S1	-1.4843304E+01 S1	-3.9254837E+00 S1
-1.7406726E-01 S0	1.7725409E-01 S0	-3.9164619E+00 S0	-1.5388185E+02 S0

AIRCRAFT STATE

MACH = .50000 ALT = 30000.00 VTRPS = 497.64
 NZ = .99307 ALPHA = 6.98570 BETA = .00000
 GAMMA = .00000 THETA = 6.98573 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 Ixz = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DP MODE: F AR MODE: F

CONTROLS

CANARD: -3.77137 FLAPS = 13.40220 STRAKES= -8.57309
 AILERON= .00000 RUDDER = .00000
 THRUST = 1896.780 % RPM = 27.33110

DYNAMICS

CXAERO = .00196910 CYAERO = .000000000 CZAERO = -.77448088
 CLMAERO= .000000000 CMMAERO= .01106507 CNMAERO= .000000000
 VC KTS = 184.311 DYN PR = 110.29778 RHO = .00089076

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.4509326	-4.990967	1.7415771	.0000000
PITCH RATE (PER RAD)	.0000000	-6.7815428	-7.4643784	.0000000
STRAKE (PER RAD)	.0368900	-.2201020	-.3320416	.0000000
MACH (PER M #)	-.0280958	-.0996858	.0370250	918.3127441
ALPHA DOT (PER RAD)	.0000000	-.4773679	-1.0246754	.0000000
CANARD (PER RAD)	-.0901213	-.2340493	.8462499	.0000000
FLAPERON (PER RAD)	.0381007	-1.4610431	-.5628066	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1382305	.2977938	-.0394745
ROLL RATE (PER RAD)	.0000000	-.0395152	-.3367432
YAW RATE (PER RAD)	.0000000	-.2018352	.2115317
AILERON (PER RAD)	-.2153511	.0800061	.2309904
RUDDER (PER RAD)	.2572526	-.1511177	.0433351

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-206.6115570	5.1300097	18.2681732
ALPHADOT (PER R/S)	-.1406762	-.0218329	.0000000
Q (PER R/S)	-2.0284662	-.1614320	.0000000
VEL (PER FT/S)	-.0822117	-.0010433	-.0035116
STRAKE (PER RAD)	-9.0755711	-.9899187	1.5211029
FLAPRN (PER RAD)	-60.2440948	-1.6779003	1.5710220
CANARD (PER RAD)	-9.6506662	2.5229330	-3.7160139

A Matrix:

B Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.3512E-02	.1827E+02	-.6052E+02	-.3194E+02	.1521E+01	.1571E+01	-.3716E+01
-.1664E-03	-.4182E+00	.9956E+00	-.7920E-02	-.1837E-01	-.1219E+00	-.1953E-01
-.1040E-02	.5139E+01	-.1832E+00	.1729E-03	-.9895E+00	-.1675E+01	.2523E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

1.9854965E+00 /+J(0.0000000E+00)
 -2.5759525E+00 /+J(0.0000000E+00)
 -7.2013699E-03 /+J(8.9473486E-02)
 -7.2013699E-03 /+J(-8.9473486E-02)

DYNAMICS

PERIOD= 70.2240 W0= .08947
 ZETA= .08023 WN= .08976

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
6.0484584E-01 S3
-5.0979300E+00 S2
-6.8874717E-02 S1
-4.1198127E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
1.5211029E+00	S3	-1.8368270E-02	S3	-9.8951769E-01	S3	0.0000000E+00	S3
6.0468521E+01	S2	-9.8885554E-01	S2	-5.1323622E-01	S2	-9.8951769E-01	S2
3.6632004E+01	S1	-6.0615614E-03	S1	-6.4046393E-03	S1	-5.1323622E-01	S1
1.64333762E+01	S0	-4.6072118E-03	S0	1.3661401E-07	S0	-6.4046383E-03	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
1.5710220E+00	S3	-1.2192947E-01	S3	-1.6752386E+00	S3	0.0000000E+00	S3
1.0010912E+02	S2	-1.6909075E+00	S2	-1.3346577E+00	S2	-1.6752386E+00	S2
9.5027222E+01	S1	-3.5190226E-03	S1	-9.4617195E-03	S1	-1.3346577E+00	S1
4.2688934E+01	S0	-4.7923326E-03	S0	6.1722756E-07	S0	-9.4617195E-03	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-3.7160139E+00	S3	-1.9532219E-02	S3	2.5233593E+00	S3	0.0000000E+00	S3
-1.5531474E+02	S2	2.5092525E+00	S2	9.6752918E-01	S2	2.5233593E+00	S2
-7.3813583E+01	S1	1.9429307E-02	S1	1.6185056E-02	S1	9.6752918E-01	S1
-3.1007690E+01	S0	1.3953928E-02	S0	-6.5585311E-08	S0	1.6185056E-02	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-9.0729885E+00	S4	-6.0226929E+01	S4	-9.6479197E+00	S4		
3.2714844E-01	S3	-7.7395020E+00	S3	-6.9680176E+00	S3		
2.4664641E+02	S2	6.5095923E+02	S2	-4.5843896E+02	S2		
-1.1204720E+00	S1	-2.9161186E+00	S1	2.6839104E+00	S1		
-2.5129035E-02	S0	-3.7328888E-02	S0	6.3363042E-02	S0		

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-46.9331818	-3.6046562	3.0218067
P (PER R/S)	.0000000	-1.1439295	-.0511982
R (PER R/S)	.0000000	.6919118	-.0342296
AILERON (PER RAD)	-8.8797045	28.9404449	1.8520613
RUDDER (PER RAD)	10.6074209	4.7141190	-1.4331026

3 Matrix:

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.9431E-01	.1216E+00	-.9926E+00	.6417E-01	-.1784E-01	.2132E-01
-.3605E+01	-.1144E+01	.6919E+00	.0000E+00	.2894E+02	.4714E+01
-.3022E+01	-.5120E-01	-.3423E-01	.0000E+00	.1852E+01	-.1433E+01
.0000E+00	.1000E+01	.1225E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.1205655E-01 +J(1.8496208E+00)
 -1.1205655E-01 +J(-1.8496208E+00)
 -1.0897570E+00 +J(0.0000000E+00)
 4.1394312E-02 +J(0.0000000E+00)

DYNAMICS

PERIOD= 3.3970 WD= 1.34962
 ZETA= .06047 WN= 1.85301

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
 1.2724695E+00 S3
 3.6234703E+00 S2
 3.5895662E+00 S1
 -1.5488935E-01 S0

NUMERATOR PHI/ AIL
 0.000000E+00 S3
 2.9167374E+01 S2
 5.1586266E+00 S1
 9.5011032E+01 S0

NUMERATOR R/ AIL
 1.8520613E+00 S3
 7.5767797E-01 S2
 1.1443089E+01 S1
 6.0404825E+00 S0

NUMERATOR P/ AIL
 2.8940445E+01 S3
 5.0657883E+00 S2
 9.3608902E+01 S1
 -7.4018013E-01 S0

NUMERATOR BETA/ AIL
 -1.7843522E-02 S3
 1.6604280E+00 S2
 1.5145521E+00 S1
 1.5081315E-01 S0

NUMERATOR PHI/ RUD
 0.000000E+00 S3
 4.535199E+00 S2
 -7.0157653E-01 S1
 9.0986328E+00 S0

NUMERATOR R/ RUD
 -1.4331026E+00 S3
 -1.9514694E+00 S2
 1.0044775E+00 S1
 5.8264428E-01 S0

NUMERATOR P/ RUD
 4.7141190E+00 S3
 -4.626195E-01 S2
 8.9755535E+00 S1
 -7.1394503E-02 S0

NUMERATOR BETA/ RUD
 2.1315321E-02 S3
 2.0209122E+00 S2
 2.0586271E+00 S1
 -6.8067074E-02 S0

NUMERATOR /

NUMERATOR AY/ RUD
 1.0528681E+01 S4
 5.0314941E+00 S3
 -6.4017953E+01 S2
 -5.8291016E+01 S1
 1.5515366E+00 S0

NUMERATOR AY/ AIL
 -8.8137894E+00 S4
 -1.6595703E+01 S3
 -1.1569971E+02 S2
 -1.0351172E+02 S1
 -5.7039185E+00 S0

AIRCRAFT STATE

MACH = .70000 ALT = 30000.00 VTRFPS = 696.70
 NZ = .99816 ALPHA = 4.42325 BETA = .00000
 GAMMA = .00000 THETA = 4.42327 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IXZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -3.02314 FLAPS = 4.89607 STRAKES= -6.07558
 AILERON= .00000 RUDDER = .00000
 THRUST = 1846.918 X RPM = 23.94859

DYNAMICS

CXAERO = -.01546481 CYAERO = .00000000 CZAERO = -.39693648
 CLMAERO= .00000000 CMMAERO= .00551222 CNMAERO= .00000000
 VC KTS = 263.097 DYN PR = 216.18355 RHO = .00089076

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LSS)
ALPHA (PER RAD)	.3493705	-5.1572371	1.7070537	.0000000
PITCH RATE(PER RAD)	.0000000	-7.2730341	-7.8922453	.0000000
STRAKE (PER RAD)	.0229956	-.2107804	-.3283005	.0000000
MACH (PER M #)	.0047504	-.1388484	.0188657	1044.1550293
ALPHA DOT (PER RAD)	.0000000	-.4738842	-1.1184721	.0000000
CANARD (PER RAD)	-.0196563	-.2862219	.8894115	.0000000
FLAPERON (PER RAD)	.0358101	-1.5469913	-.5789517	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1047325	.2481541	-.0427122
ROLL RATE (PER RAD)	.0000000	-.0224129	-.3665565
YAW RATE (PER RAD)	.0000000	-.1851723	.1751904
AILERON (PER RAD)	-.2181808	.0766279	.2116327
RUDDER (PER RAD)	.2568131	-.1514564	.0444041

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-419.8540039	9.9265556	27.8958130
ALPHADOT (PER R/S)	-1972643	-0.0336635	.0000000
Q (PER R/S)	-3.0456676	-2389598	.0000000
VEL (PER FT/S)	-0569184	-0009199	-0041988
STRAKE (PER RAD)	-17.0347595	-1.9183798	1.8584509
FLAPRN (PER RAD)	-125.0240021	-3.3830261	2.8940830
CANARD (PER RAD)	-23.1317596	5.1971560	-1.5685735

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-4199E-02	.2790E+02	-.5373E+02	-.3208E+02	-1858E+01	.2894E+01	-.1589E+01
-8192E-04	-.6043E+00	.9953E+00	-.3571E-02	-.2452E-01	-.1799E+00	-.3329E-01
-.9171E-03	.9947E+01	-.2725E+00	-1202E-03	-.1918E+01	-.3377E+01	.5198E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-3.5954370E+00	+J(0.0000000E+00)
2.7233095E+00	+J(0.0000000E+00)
-4.4075027E-03	+J(6.6805065E-02)
-4.4075027E-03	+J(-6.6805065E-02)

DYNAMICS

PERIOD= 94.0525	WD= .06681
ZETA= .06583	WN= .06695

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
8.8092566E-01 S3
-9.7792645E+00 S2
-8.2319975E-02 S1
-4.3853033E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
1.8584509E+00 S3		-2.4516709E-02 S3		-1.9175549E+00 S3		0.0000000E+00 S3	
1.0397913E+02 S2		-1.9155407E+00 S2		-1.4123249E+00 S2		-1.9175549E+00 S2	
6.5351959E+01 S1		-1.0160878E-02 S1		-1.2186185E-02 S1		-1.4123249E+00 S1	
4.5248718E+01 S0		-4.2810515E-03 S0		5.2395496E-07 S0		-1.2186185E-02 S0	
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
2.5940830E+00 S3		-1.7993659E-01 S3		-3.3769684E+00 S3		0.0000000E+00 S3	
1.7896846E+02 S2		-3.4112272E+00 S2		-3.8472137E+00 S2		-3.3769684E+00 S2	
1.9083232E+02 S1		-1.0940619E-02 S1		-2.3155238E-02 S1		-3.8472137E+00 S1	
1.2331104E+02 S0		-3.5171481E-03 S0		2.1631395E-06 S0		-2.3155238E-02 S0	
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-1.5885735E+00 S3		-3.3291612E-02 S3		5.1982765E+00 S3		0.0000000E+00 S3	
-2.8163428E+02 S2		5.1649342E+00 S2		2.832491E+00 S2		5.1982765E+00 S2	
-1.5818919E+02 S1		2.9132813E-02 S1		2.6698925E-02 S1		2.832491E+00 S1	
-9.0712402E+01 S0		1.4551431E-02 S0		-5.1193530E-07 S0		2.6698925E-02 S0	
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-1.7029922E+01 S4		-1.2498849E+02 S4		-2.3125198E+01 S4			
1.3989258E+00 S3		-2.3797119E+01 S3		-2.3160400E+01 S3			
9.6922046E+02 S2		2.6563928E+03 S2		-1.9349111E+03 S2			
1.9866123E+00 S1		4.0947742E+00 S1		-1.4076309E+00 S1			
-3.0602247E-02 S0		-5.8959149E-02 S0		6.6605389E-02 S0			

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y	L	N
	FT/SEC**2	RAD/SEC**2	RAD/SEC**2
BETA (PER RAD)	-89.2818604	-8.2696390	4.8526182
P (PER R/S)	.0000000	-1.7398958	-.0695753
R (PER R/S)	.0000000	.7992898	-.0470750
AILERON (PER RAD)	-17.6328583	51.9977722	3.3957148
RUDDER (PER RAD)	20.7350354	9.4961958	-2.8069754

z Matrix:

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.1281E+00	.7712E-01	-.9970E+00	.4604E-01	-.2531E-01	.2979E-01
-.8270E+01	-.1740E+01	.7993E+00	.0000E+00	.5200E+02	.9496E+01
.4853E+01	-.6958E-01	-.4707E-01	.0000E+00	.3396E+01	-.2807E+01
.0000E+00	.1000E+01	.7735E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.2383652E-01 +J(2.3344107E+00)
 -1.2383652E-01 +J(-2.3344107E+00)
 -1.6883583E+00 +J(0.0000000E+00)
 2.0894349E-02 +J(0.0000000E+00)

DYNAMICS

PERIOD= 2.6916 WD= 2.33441
 ZETA= .05297 WN= 2.33769

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 1.9151201E+00 S3
 5.8424635E+00 S2
 9.1035395E+00 S1
 -1.9278216E-01 S0

NUMERATOR PHI/ AIL
 0.0000000E+00 S3
 5.2260437E+01 S2
 1.2236065E+01 S1
 2.8182251E+02 S0

NUMERATOR R/ AIL
 3.3957148E+00 S3
 2.6027746E+00 S2
 2.1691162E+01 S1
 1.2910760E+01 S0

NUMERATOR PHI/ RUD
 0.0000000E+00 S3
 9.2790651E+00 S2
 -1.2715130E+00 S1
 2.2776596E+01 S0

NUMERATOR R/ RUD
 -2.8069754E+00 S3
 -5.7596960E+00 S2
 1.3218460E+00 S1
 1.0529442E+00 S0

NUMERATOR P/ AIL
 5.1997772E+01 S3
 1.2034730E+01 S2
 2.8014453E+02 S1
 -9.9877137E-01 S0

NUMERATOR P/ RUD
 9.4961958E+00 S3
 -8.2597911E-01 S2
 2.2674347E+01 S1
 -8.1455231E-02 S0

NUMERATOR AY/ RUD
 2.0693222E+01 S4
 2.9651855E+01 S3
 -2.1082544E+02 S2
 -3.3053516E+02 S1
 5.1470003E+00 S0

NUMERATOR BETA/ AIL
 -2.5309104E-02 S3
 5.7941401E-01 S2
 5.1724392E-01 S1
 2.4582928E-01 S0

NUMERATOR BETA/ RUD
 2.9790483E-02 S3
 3.5842285E+00 S2
 5.8208065E+00 S1
 -1.0246772E-01 S0

NUMERATOR AY/ AIL
 -1.7580338E+01 S4
 -3.2709717E+01 S3
 -1.558324E+02 S2
 -2.0719141E+02 S1
 -1.8545135E+01 S0

AIRCRAFT STATE

MACH = .90000 ALT = 30000.00 VTRFPS = 895.76
 NZ = .99930 ALPHA = 3.80104 BETA = .00000
 GAMMA = .00000 THETA = 3.80106 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IXZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -3.85964 FLAPS = -.46384 STRAKES= -4.60731
 AILERON= .00000 RUDDER = .00000
 THRUST = 2441.760 X RPM = 28.30701

DYNAMICS

CXAERO = -.02095872 CYAERO = .00000000 CZAERO = -.24029338
 CLMAERO= .00000000 CMMAERO= .00440415 CNMAERO= .00000000
 VC KTS = 346.571 DYN PR = 357.36450 RHO = .00089076

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.3193833	-5.9415541	1.7237587	.0000000
PITCH RATE (PER RAD)	.0000000	-8.0019798	-8.8417330	.0000000
STRAKE (PER RAD)	.0093769	-.1897161	-.2548249	.0000000
MACH (PER M #)	-.0749215	-.2726399	-.1337801	1353.0637207
ALPHA DOT (PER RAD)	.0000000	-.4482737	-1.2793598	.0000000
CANARD (PER RAD)	-.0062467	-.4191260	1.0069675	.0000000
FLAPERON (PER RAD)	.0309398	-1.5183449	-.6590127	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.0788183	.2237321	-.0706861
ROLL RATE (PER RAD)	.0000000	-.0116976	-.4041967
YAW RATE (PER RAD)	.0000000	-.1601692	.1834191
AILERON (PER RAD)	-.2980881	.0524364	.1603577
RUDDER (PER RAD)	.2385777	-.1413351	.0447574

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-796.6970215	16.5060730	41.5146484
ALPHADOT (PER R/S)	-.2402936	-.0495848	-.0000000
Q (PER R/S)	-4.3083305	-.3441961	-.0000000
VEL (PER FT/S)	-.0492910	-.0024566	-.0166839
STRAKE (PER RAD)	-25.3453827	-2.4614649	1.3195210
FLAPRN (PER RAD)	-202.8453979	-6.3656931	4.1334457
CANARD (PER RAD)	-55.9936981	9.7267437	-.8345309

A Matrix:

B Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.1668E-01	.4151E+02	-.5938E+02	-.3210E+02	.1320E+01	.4133E+01	-.8345E+00
-.5513E-04	-.8911E+00	.9949E+00	-.2386E-02	-.2835E-01	-.2269E+00	-.6263E-01
-.2454E-02	.1655E+02	-.3935E+00	.1183E-03	-.2460E+01	-.6354E+01	.9730E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-4.7228603E+00 +J(0.0000000E+00)
3.4459734E+00 +J(0.0000000E+00)
-1.2221541E-02 +J(7.7060878E-02)
-1.2221541E-02 +J(-7.7060878E-02)

DYNAMICS

PERIOD= 81.5353 WD= -.07706
ZETA= -.15664 WN= .07802

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
1.3013449E+00 S3
-1.6237488E+01 S2
-3.9004117E-01 S1
-9.9083841E-02 S0

NUMERATOR U/STRK
1.3195210E+00 S3
1.4659949E+02 S2
1.1367815E+02 S1
8.5736298E+01 S0

NUMERATOR U/FLAP
4.1334457E+00 S3
3.7322510E+02 S2
4.3045459E+02 S1
3.0313257E+02 S0

NUMERATOR U/CAN
-8.3453089E-01 S3
-5.8144263E+02 S2
-3.5137549E+02 S1
-2.4607307E+02 S0

NUMERATOR AN/STRK
-2.5338577E+01 S4
7.2705078E-01 S3
2.3752600E+03 S2
3.8219543E+01 S1
-1.1844641E-01 S0

NUMERATOR ALP/STRK
-2.8349683E-02 S3
-2.4592457E+00 S2
-4.2319588E-02 S1
-2.0153148E-03 S0

NUMERATOR ALP/FLAP
-2.2688955E-01 S3
-6.4154148E+00 S2
-8.9699149E-02 S1
6.9046989E-03 S0

NUMERATOR ALP/CAN
-6.2630892E-02 S3
9.6547003E+00 S2
1.8091846E-01 S1
2.1764699E-02 S0

NUMERATOR AN/FLAP
-2.0279089E+02 S4
-5.4496094E+01 S3
8.4275391E+03 S2
1.3005711E+02 S1
-3.6760068E-01 S0

NUMERATOR Q/STRK
-2.4600592E+00 S3
-2.7057142E+00 S2
-5.1233392E-02 S1
1.0262418E-05 S0

NUMERATOR Q/FLAP
-6.3544426E+00 S3
-9.5338907E+00 S2
-1.6135871E-01 S1
2.6230060E-05 S0

NUMERATOR Q/CAN
9.7298439E+00 S3
7.7984037E+00 S2
1.5859735E-01 S1
-4.0898129E-05 S0

NUMERATOR AN/CAN
-5.5978668E+01 S4
-6.7164063E+01 S3
-6.7876523E+03 S2
-1.0566628E+02 S1
3.7482125E-01 S0

NUMERATOR THT/STRK
0.0000000E+00 S3
-2.4600592E+00 S2
-2.7057142E+00 S1
-5.1233392E-02 S0

NUMERATOR THT/FLAP
0.0000000E+00 S3
-6.3544426E+00 S2
-9.5338907E+00 S1
-1.6135871E-01 S0

NUMERATOR THT/CAN
0.0000000E+00 S3
9.7298439E+00 S2
7.7984037E+00 S1
1.5859735E-01 S0

LATERAL-DIR. SYSTEM
BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-144.1262207	-25.2312775	6.7833357
P (PER R/S)	.0000000	-2.4639597	-.0917765
R (PER R/S)	.0000000	1.0830641	-.0455994
AILERON (PER RAD)	-39.8235016	65.0513458	4.0579405
RUDDER (PER RAD)	31.8731232	15.9806738	-4.2836924

B Matrix:

	P	R	PHI	AIL	RUD
BETA					
-1609E+00	.6629E-01	-.9978E+00	.3584E-01	-.4446E-01	.3558E-01
-.2523E+02	-.2464E+01	.1083E+01	.0000E+00	.6505E+02	.1598E+02
.6783E+01	-.9178E-01	-.4560E-01	.0000E+00	.4058E+01	-.4284E+01
.0000E+00	.1000E+01	.6644E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.6325730E-01 +J(2.8804131E+00)
-1.6325730E-01 +J(-2.8804131E+00)
-2.3575726E+00 +J(0.0000000E+00)
1.3626251E-02 +J(0.0000000E+00)

DYNAMICS

PERIOD= 2.1813 Wd= 2.88041
ZETA= .05659 Wn= 2.88504

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
2.6704578E+00 S3
9.0565758E+00 S2
1.9499069E+01 S1
-2.6738268E-01 S0

NUMERATOR PHI/ AIL
0.000000E+00 S3
6.5320938E+01 S2
1.9240692E+01 S1
5.4574634E+02 S0

NUMERATOR R/ AIL
4.0579405E+00 S3
4.3797617E+00 S2
3.5841736E+01 S1
1.9484039E+01 S0

NUMERATOR P/ AIL
6.5051346E+01 S3
1.8949707E+01 S2
5.4336499E+02 S1
-1.2945642E+00 S0

NUMERATOR BETA/ AIL
-4.4457916E-02 S3
1.5177602E-01 S2
-1.1999264E+00 S1
2.7341551E-01 S0

NUMERATOR PHI/ RUD
0.000000E+00 S3
1.5696074E+01 S2
-3.0657587E+00 S1
-1.7234641E-01 S0

NUMERATOR R/ RUD
-4.2836924E+00 S3
-1.2469370E+01 S2
-1.2359648E+00 S1
1.1451665E-02 S0

NUMERATOR P/ RUD
1.5980674E+01 S3
-2.2373219E+00 S2
-9.0231717E-02 S1
-7.7521522E-04 S0

NUMERATOR BETA/ RUD
3.5582323E-02 S3
5.4229499E+00 S2
1.2303866E+01 S1
-1.6878706E-01 S0

NUMERATOR /

NUMERATOR AY/ RUD
3.1803009E+01 S4
6.9301270E+01 S3
-5.1718018E+02 S2
-1.1517698E+03 S1
1.5814255E+01 S0

NUMERATOR AY/ AIL
-3.9735901E+01 S4
-1.0023828E+02 S3
-3.8017407E+02 S2
-6.0411328E+02 S1
-2.8767838E+01 S0

AIRCRAFT STATE

MACH = 1.10000 ALT = 30000.00 VTRFPS = 1094.81
 NZ = .99930 ALPHA = 3.06149 BETA = .00000
 GAMMA = .00000 THETA = 3.06151 ACOSW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IYZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.77283 FLAPS = -1.18431 STRAKES= -7.48768
 AILERON= .00000 RUDDER = .00000
 THRUST = 6109.574 X RPM = 60.54581

DYNAMICS

CXAERO = -.05323606 CYAERO = .00000000 CZAERO = -.16099501
 CLMAERO= .00000000 CMMAERO= .00737481 CNMAERO= .00000000
 VC KTS = 435.123 DYN PR = 533.84131 RHO = .00089076

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2543105	-6.2660198	-8387392	.0000000
PITCH RATE(PER RAD)	.0000000	-7.0107250	-9.1285238	.0000000
STRAKE (PER RAD)	.0108321	-.1586864	-.2843701	.0000000
MACH (PER M #)	-.0455308	.2147942	.0232477	2621.9914551
ALPHA DOT (PER RAD)	.0000000	-.4070342	-1.5345993	.0000000
CANARD (PER RAD)	-.0217344	-.6669040	.8720989	.0000000
FLAPERON (PER RAD)	.0278860	-.8441584	-.6222041	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1677351	.2246048	-.0842797
ROLL RATE (PER RAD)	.0000000	.0644804	-.4943840
YAW RATE (PER RAD)	.0000000	-.4039204	.2384676
AILERON (PER RAD)	-.3552630	.0756314	.1011095
RUDDER (PER RAD)	.1345730	-.0834191	.0268597

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-1247.8579102	12.0991507	48.9418182
ALPHADOT (PER R/S)	-.2670870	-.0728071	.0000000
Q (PER R/S)	-4.6134424	-.4343302	.0000000
VEL (PER FT/S)	.0454000	-.0001056	-.0256512
STRAKE (PER RAD)	-31.6691132	-4.1033373	2.1617632
FLAPRN (PER RAD)	-168.4690399	-8.9781361	5.5652122
CANARD (PER RAD)	-133.0943146	12.5840082	-4.3375406

A Matrix:

B Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.2565E-01	.4894E+02	-.5847E+02	-.3213E+02	.2162E+01	.5565E+01	-.4338E+01
.4152E-04	-.1141E+01	.9955E+00	-.1571E-02	-.2896E-01	-.1541E+00	-.1217E+00
-.1086E-03	.1218E+02	-.5068E+00	.1144E-03	-.4101E+01	-.8967E+01	.1259E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-4.3234386E+00	+J(0.0000000E+00)
-4.5898989E-02	+J(0.0000000E+00)
2.671681E+00	+J(0.0000000E+00)
2.4034203E-02	+J(0.0000000E+00)

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
1.6736021E+00 S3
-1.1515745E+01 S2
-2.5414598E-01 S1
1.2747411E-02 S0

NUMERATOR THT/STRK
0.0000000E+00 S3
-4.1012287E+00 S2
-5.1383133E+00 S1
-1.1978424E-01 S0

NUMERATOR Q/STRK
-4.1012287E+00 S3
-5.1383133E+00 S2
-1.1978424E-01 S1
7.8580315E-06 S0

NUMERATOR ALP/STRK
-2.8967705E-02 S3
-4.0982561E+00 S2
-8.8706970E-02 S1
5.7373755E-03 S0

NUMERATOR THT/FLAP
0.0000000E+00 S3
-8.9669189E+00 S2
-1.2339924E+01 S1
-2.8944868E-01 S0

NUMERATOR Q/FLAP
-8.9669189E+00 S3
-1.2339924E+01 S2
-2.8944868E-01 S1
1.7188941E-05 S0

NUMERATOR ALP/FLAP
-1.5406120E-01 S3
-9.0087004E+00 S2
-1.9461489E-01 S1
1.2861323E-02 S0

NUMERATOR THT/CAN
0.0000000E+00 S3
1.2592870E+01 S2
1.3210980E+01 S1
3.0397624E-01 S0

NUMERATOR Q/CAN
1.2592870E+01 S3
1.3210980E+01 S2
3.0397624E-01 S1
-2.4109293E-05 S0

NUMERATOR ALP/CAN
-1.2171179E-01 S3
1.2471680E+01 S2
2.7079856E-01 S1
-1.6880456E-02 S0

NUMERATOR AN/STRK
-3.1661377E+01 S4
3.2500000E+00 S3
5.5134492E+03 S2
1.2839757E+02 S1
-2.1441990E-01 S0

NUMERATOR AN/CAN
-1.3306180E+02 S4
-1.3249219E+02 S3
-1.4125254E+04 S2
-3.2807666E+02 S1
5.4868954E-01 S0

NUMERATOR AN/FLAP
-1.6842789E+02 S4
-4.5675781E+01 S3
1.3262492E+04 S2
3.0929736E+02 S1
-5.1616073E-01 S0

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-233.0455017	-45.8180847	9.8951597
P (PER R/S)	.0000000	-3.6630659	-.0864285
R (PER R/S)	.0000000	1.6703815	-.1985572
AILERON (PER RAD)	-70.9000549	62.1588593	6.0304470
RUDDER (PER RAD)	26.8568115	14.3553901	-3.7676687

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.2129E+00	.5341E-01	-.9986E+00	.2935E-01	-.6476E-01	.2453E-01
-.4582E+02	-.3663E+01	.1670E+01	.0000E+00	-.6216E+02	.1436E+02
-.9895E+01	-.8643E-01	-.1986E+00	.0000E+00	.6030E+01	-.3768E+01
.0000E+00	.1000E+01	.5348E-01	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-3.0630112E-01 +J(-3.4396019E+00)
-3.0630112E-01 +J(-3.4396019E+00)
-3.4686584E+00 +J(0.0000000E+00)
6.7980587E-03 +J(0.0000000E+00)

DYNAMICS

PERIOD= 1.8267 WD= 3.43960
ZETA= .08870 WN= 3.45321

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 4.0744858E+00 S3
 1.4021755E+01 S2
 4.1266907E+01 S1
 -2.8119493E-01 S0

NUMERATOR PHI/ AIL
 0.0000000E+00 S3
 6.2481384E+01 S2
 3.9542206E+01 S1
 8.9699023E+02 S0

NUMERATOR R/ AIL
 6.0304470E+00 S3
 1.7360474E+01 S2
 4.8560833E+01 S1
 2.6157883E+01 S0

NUMERATOR P/ AIL
 6.2158959E+01 S3
 3.8613708E+01 S2
 8.9439307E+02 S1
 -1.3998194E+00 S0

NUMERATOR BETA/ AIL
 -6.4759831E-02 S3
 -2.9521704E+00 S2
 -1.371952E+01 S1
 6.8403101E-01 S0

NUMERATOR PHI/ RUD
 0.0000000E+00 S3
 1.4153890E+01 S2
 -2.3457184E+00 S1
 -3.1291214E+01 S0

NUMERATOR R/ RUD
 -3.7676697E+00 S3
 -1.5601193E+01 S2
 -3.8486757E+00 S1
 -8.9729071E-01 S0

NUMERATOR P/ RUD
 1.4355399E+01 S3
 -1.5113068E+00 S2
 -3.1085373E+01 S1
 4.7830295E-02 S0

NUMERATOR BETA/ RUD
 2.4530895E-02 S3
 4.6237049E+00 S2
 1.5273315E+01 S1
 -1.2465143E-01 S0

NUMERATOR /

NUMERATOR AY/ RUD
 2.6818481E+01 S4
 9.6483887E+01 S3
 -7.2482324E+02 S2
 -2.4508887E+03 S1
 2.1563889E+01 S0

NUMERATOR AY/ AIL
 -7.0798920E+01 S4
 -2.6917700E+02 S3
 2.8471729E+02 S2
 2.7038281E+02 S1
 -1.3953349E+02 S0

AIRCRAFT STATE

MACH =	1.30000	ALT =	30000.00	VTRFPS =	1293.87
NZ =	.99880	ALPHA =	2.90715	BETA =	.00000
GAMMA =	.00000	THETA =	2.90717	ACGW =	15926.00
IXX =	4548.00	IYY =	49429.00	IZZ =	52531.00
IXZ =	1827.00	XCG =	450.56006	ZCG =	64.88000
GEAR DOWN:	F	NY =	.00000	NX =	.00000
NORMAL MODE:	T	DR MODE:	F	AR MODE:	F

CONTROLS

CANARDS=	-1.98113	FLAPS =	-1.68678	STRAKES=	-8.75000
AILERON=	.00000	RUDDER =	.00000		
THRUST =	9583.355	% RPM =	87.75790		

DYNAMICS

CXAERO =	-.06360304	CYAERO =	.00000000	CZAERO =	-.11527956
CLMAERO=	.00000000	CMMAERO=	.00827518	CNMAERO=	.00000000
VC KTS =	522.812	DYN PR =	745.61182	RHO =	.00089076

A/C CONSTANTS

AREA =	185.0490	CBAR =	7.2200	SPAN =	27.2000
--------	----------	--------	--------	--------	---------

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2021586	-5.6125793	.3250694	.0000000
PITCH RATE(PER RAD)	.0000000	-5.6427326	-7.8051939	.0000000
STRAKE (PER RAD)	.0097723	-.1229883	-.2577143	.0000000
MACH (PER M #)	-.0254503	.0542617	-.1931579	5525.8203125
ALPHA DOT (PER RAD)	.0000000	-.3696541	-1.3849792	.0000000
CANARD (PER RAD)	-.0276658	-.5066074	.6671954	.0000000
FLAPERON (PER RAD)	.0200535	-.5758265	-.6038276	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.2073336	.2343938	-.0891617
ROLL RATE (PER RAD)	.0000000	.0640213	-.5135223
YAW RATE (PER RAD)	.0000000	-.4308162	.2139636
AILERON (PER RAD)	-.2677095	.0993752	.0615668
RUDDER (PER RAD)	.0717860	-.0524813	.0153248

LONGITUDINAL SYSTEM
 90DY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-1562.6762695	6.2950411	53.9412994
ALPHADOT (PER R/S)	-.2867408	-.0776773	.06J0000
Q (PER R/S)	-4.3883591	-.4388876	.0000000
VEL (PER FT/S)	.0268946	-.0040022	-.0254798
STRAKE (PER RAD)	-34.2815552	-5.1938839	2.7239237
FLAPRN (PER RAD)	-160.5049438	-12.1693316	5.5996845
CANARD (PER RAD)	-141.2109222	13.4464226	-7.7115173

B Matrix:

	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
VEL						
-2548E-01	.5394E+02	-.6562E+02	-.3213E+02	.2724E+01	.5590E+01	-.7712E+01
.2081E-04	-.1209E+01	.9964E+00	-.1263E-02	-.2652E-01	-.1242E+00	-.1093E+00
-.4004E-02	.6389E+01	-.5163E+00	.9807E-04	-.5192E+01	-.1216E+02	.1345E+02
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-3.4514618E+00	,+J(0.0000000E+00)
1.7667542E+00	,+J(0.0000000E+00)
-3.3055067E-02	,+J(1.5403634E-01)
-3.3055067E-02	,+J(-1.5403634E-01)

DYNAMICS

PERIOD= 40.7903	WD= .15404
ZETA= .20982	WN= .15754

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 1.7508039E+00 S3
 -5.9616385E+00 S2
 -3.6132950E-01 S1
 -1.5134710E-01 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
2.7239237E+00 S3		-2.6523590E-02 S3		-5.1918240E+00 S3		0.0000000E+00 S3	
3.4396533E+02 S2		-5.1873589E+00 S2		-6.5897751E+00 S2		-5.1918240E+00 S2	
2.9444312E+02 S1		-1.2237698E-01 S1		-1.6552413E-01 S1		-6.5897751E+00 S1	
2.0732071E+02 S0		7.0635192E-03 S0		4.4522394E-06 S0		-1.6552413E-01 S0	
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
5.5896845E+00 S3		-1.2418246E-01 S3		-1.2159685E+01 S3		0.0000000E+00 S3	
8.0088525E+02 S2		-1.2182865E+01 S2		-1.5827151E+01 S2		-1.2159685E+01 S2	
7.1843774E+02 S1		-2.6797819E-01 S1		-3.8065135E-01 S1		-1.5827151E+01 S1	
4.9876563E+02 S0		2.4524089E-02 S0		1.0337829E-05 S0		-3.8065135E-01 S0	
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-7.7115173E+00 S3		-1.0925466E-01 S3		1.3454909E+01 S3		0.0000000E+00 S3	
-9.0213306E+02 S2		1.3346892E+01 S2		1.5943207E+01 S2		1.3454909E+01 S2	
-6.8965405E+02 S1		3.6418360E-01 S1		4.4150287E-01 S1		1.5943207E+01 S1	
-5.0126563E+02 S0		4.5911595E-03 S0		-1.1793746E-05 S0		4.4150287E-01 S0	
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-3.4273956E+01 S4		-1.6046939E+02 S4		-1.4117958E+02 S4			
5.7695313E+00 S3		-2.9953125E+01 S3		-1.3958203E+02 S3			
8.3487422E+03 S2		2.0085824E+04 S2		-2.0109359E+04 S2			
2.1226592E+02 S1		4.9774390E+02 S1		-5.3856445E+02 S1			
-2.7585214E-01 S0		-6.3449776E-01 S0		7.3567545E-01 S0			

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-336.5300293	-67.7944031	14.3876352
P (PER R/S)	.0000000	-4.4975595	-1083472
R (PER R/S)	.0000000	1.7503071	-2626379
AILERON (PER RAD)	-74.6208801	54.4156952	8.9920788
RUDDER (PER RAD)	20.0095062	11.2973680	-3.3564367

9 Matrix:

A Matrix:

	P	R	PHI	AIL	RUD
BETA	.5072E-01	-.9987E+00	.2483E-01	-.5767E-01	.1546E-01
P	-.2601E+00	-.4498E+01	.1750E+01	.5442E+02	.1130E+02
R	-.6779E+02	-.1083E+00	.0000E+00	.8992E+01	-.3336E+01
AILERON	.1439E+02	-.2626E+00	.0000E+00	.0000E+00	.0000E+00
RUDDER	.0000E+00	.1000E+01	.5078E-01	.0000E+00	.0000E+00

SYSTEM POLES

-3.7424850E-01	,+J(4.1345549E+00)
-3.7424850E-01	,+J(-4.1345549E+00)
-4.2754822E+00	,+J(0.0000000E+00)
3.7193897E-03	,+J(0.0000000E+00)

DYNAMICS

PERIOD= 1.5197	WD= 4.13435
ZETA= .09015	WN= 4.15146

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
5.0202923E+00 S3
2.0416443E+01 S2
7.3609711E+01 S1
-2.7412945E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-5.7672545E-02 S3	5.4415695E+01 S3	8.9920788E+00 S3	0.000000E+00 S3
-6.4952192E+00 S2	4.8093628E+01 S2	3.6055634E+01 S2	5.4872330E+01 S2
-3.1695435E+01 S1	1.3981187E+03 S1	7.5455032E+01 S1	4.9924637E+01 S1
7.8938073E-01 S0	-1.7573566E+00 S0	3.4582474E+01 S0	1.4019504E+03 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
1.5464831E-02 S3	1.1297368E+01 S3	-3.3564367E+00 S3	0.000000E+00 S3
3.9987049E+00 S2	-1.0177202E+00 S2	-1.6970306E+01 S2	1.1126919E+01 S2
1.6448853E+01 S1	-6.5563751E+01 S1	-6.4272776E+00 S1	-1.8795185E+00 S1
-9.2803359E-02 S0	8.1937850E-02 S0	-1.6143332E+00 S0	-6.5890137E+01 S0

NUMERATOR AY/ AIL	NUMERATOR AY/ RUD
-7.4524841E+01 S4	1.9983749E+01 S4
-3.6439600E+02 S3	8.8588379E+01 S3
7.1509790E+02 S2	-9.6457471E+02 S2
5.1723125E+03 S1	-4.0624922E+03 S1
-2.4527293E+02 S0	2.5791794E+01 S0

AIRCRAFT STATE

MACH = .70000 ALT = 40000.00 VTRFPS = 677.95
 NZ = .99592 ALPHA = 5.68030 BETA = .00000
 GAMMA = .00000 THETA = 5.68032 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 IXZ = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -2.77892 FLAPS = 9.99677 STRAKES= -7.26573
 AILERON= .00300 RUDDER = .00000
 THRUST = 1840.380 % RPM = 38.72850

DYNAMICS

CYAERO = -.01057255 CYAERO = .00000000 CZAERO = -.63449287
 CLMAERO= .00000000 CMMAERO= .00877998 CNMAERO= .00000000
 VC KTS = 209.378 DYN PR = 134.97624 RHO = .00058734

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.3641798	-5.2396326	1.6744871	.0000000
PITCH RATE (PER RAD)	.0000000	-7.2729263	-7.8916559	.0000000
STRAKE (PER RAD)	.0262024	-2.2128500	-.3325030	.0000000
MACH (PER M #)	-.0316767	-.2483886	.0187909	1324.5139160
ALPHA DOT (PER RAD)	.0000000	-.4739590	-1.1184778	.0000000
CANARD (PER RAD)	-.0605030	-.3027340	.8751573	.0000000
FLAPERON (PER RAD)	.0358103	-1.5469923	-.5789520	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1372070	.2638459	-.0358600
ROLL RATE (PER RAD)	.0000000	-.0337078	-.3552643
YAW RATE (PER RAD)	.0000000	-.1845067	.2046546
AILERON (PER RAD)	-.2182385	.0754722	.2177144
RUDDER (PER RAD)	.2681669	-.1566859	.0460607

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-268.9685059	6.0603180	17.9809265
ALPHADOT (PER R/S)	-.1260996	-.0215158	.0000000
Q (PER R/S)	-1.9541512	-.1533111	.0000000
VEL (PER FT/S)	-.0682644	-.0007512	-.0031283
STRAKE (PER RAD)	-10.7402487	-1.2130899	1.3221540
FLAPRN (PER RAD)	-78.0600586	-2.1122236	1.8069592
CANARD (PER RAD)	-15.2757215	3.1928873	-3.0529356

B Matrix:

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.3128E-02	.1798E+02	-.6710E+02	-.3202E+02	.1322E+01	.1807E+01	-.3053E+01
-.1012E-03	-.3986E+00	.9969E+00	-.4720E-02	-.1592E-01	-.1157E+00	-.2264E-01
-.7491E-03	.6069E+01	-.1748E+00	.1015E-03	-.1213E+01	-.2110E+01	.3193E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-2.7564287E+00 ,+J(0.0000000E+00)
2.1905737E+00 ,+J(0.0000000E+00)
-5.3282194E-03 ,+J(6.9342077E-02)
-5.3282194E-03 ,+J(-6.9342077E-02)

DYNAMICS

PERIOD= 90.6114 WD= .06934
ZETA= .07661 WN= .06955

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 5.7650864E-01 S3
 -6.0272713E+00 S2
 -6.1579615E-02 S1
 -2.9190347E-02 S0

NUMERATOR THT/STRK
 0.0000000E+00 S3
 -1.2127476E+00 S2
 -5.8481032E-01 S1
 -5.0124153E-03 S0

NUMERATOR Q/STRK
 -1.2127476E+00 S3
 -5.8481032E-01 S2
 -5.0124153E-03 S1
 1.5358103E-07 S0

NUMERATOR ALP/STRK
 -1.5917383E-02 S3
 -1.2119741E+00 S2
 -6.5092221E-03 S1
 -3.5234375E-03 S0

NUMERATOR U/STRK
 1.3221540E+00 S3
 8.1849289E+01 S2
 4.8051651E+01 S1
 1.8710846E+01 S0

NUMERATOR THT/FLAP
 0.0000000E+00 S3
 -2.1097345E+00 S2
 -1.5510302E+00 S1
 -8.7552108E-03 S0

NUMERATOR Q/FLAP
 -2.1097345E+00 S3
 -1.5510302E+00 S2
 -8.7552108E-03 S1
 4.4321797E-07 S0

NUMERATOR ALP/FLAP
 -1.1568743E-01 S3
 -2.123929E+00 S2
 -6.5628896E-03 S1
 -4.0208884E-03 S0

NUMERATOR U/FLAP
 1.8069592E+00 S3
 1.4052264E+02 S2
 1.2210005E+02 S1
 4.9634048E+01 S0

NUMERATOR THT/CAN
 0.0000000E+00 S3
 3.1933746E+00 S2
 1.1478243E+00 S1
 1.2451384E-02 S0

NUMERATOR Q/CAN
 3.1933746E+00 S3
 1.1478243E+00 S2
 1.2451384E-02 S1
 -2.7503677E-07 S0

NUMERATOR ALP/CAN
 -2.2639096E-02 S3
 3.1798096E+00 S2
 2.0028185E-02 S1
 1.0827433E-02 S0

NUMERATOR U/CAN
 -3.0529356E+00 S3
 -2.1643839E+02 S2
 -1.0300623E+02 S1
 -3.6713929E+01 S0

NUMERATOR AN/CAN
 -1.5272961E+01 S4
 -9.1513672E+00 S3
 -7.5066821E+02 S2
 2.5596762E+00 S1
 3.9836809E-02 S0

NUMERATOR AN/FLAP
 -7.8045441E+01 S4
 -9.6188965E+00 S3
 1.0352151E+03 S2
 -1.7453490E+00 S1
 -2.8179854E-02 S0

NUMERATOR AN/STRK
 -1.0738237E+01 S4
 5.2172852E-01 S3
 3.8627319E+02 S2
 -8.5732146E-01 S1
 -1.6065575E-02 S0

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-57.3826141	-4.0425940	3.2716923
P (PER R/S)	.0000000	-1.0832376	-.0464195
R (PER R/S)	.0000000	.6024585	-.0269151
AILERON (PER RAD)	-11.0121489	33.3805389	2.1370306
RUDDER (PER RAD)	13.5314951	6.1524420	-1.8124237

B Matrix:

A Matrix:

BETA	P	R	PHI	AIL	RUD
-.8464E-01	.9898E-01	-.9951E+00	.4722E-01	-.1624E-01	.1996E-01
-.4043E+01	-.1083E+01	.6025E+00	.0000E+00	.3338E+02	.6152E+01
.3272E+01	-.4642E-01	-.2692E-01	.0000E+00	.2137E+01	-.1812E+01
.0000E+00	.1000E+01	.9947E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-8.9169919E-02 +J(1.9095812E+00)
 -8.9169919E-02 +J(-1.9095812E+00)
 -1.0441036E+00 +J(0.0000000E+00)
 2.7642813E-02 +J(0.0000000E+00)

DYNAMICS

PERIOD= 3.2903 WD= 1.90958
 ZETA= .04664 WN= 1.91166

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 1.1947937E+00 S3
 3.8068371E+00 S2
 3.7094078E+00 S1
 -1.0547316E-01 S0

NUMERATOR PHI/ AIL
 0.0000000E+00 S3
 3.3593094E+01 S2
 5.1657743E+00 S1
 1.1858672E+02 S0

NUMERATOR R/ AIL
 2.1370306E+00 S3
 8.9314073E-01 S2
 1.1668612E+01 S1
 5.5654163E+00 S0

NUMERATOR P/ AIL
 3.3380539E+01 S3
 5.0769377E+00 S2
 1.1742609E+02 S1
 -5.5359797E-01 S0

NUMERATOR BETA/ AIL
 -1.6243253E-02 S3
 1.1593361E+00 S2
 1.0402021E+00 S1
 1.0682392E-01 S0

NUMERATOR PHI/ RUD
 0.0000000E+00 S3
 5.9721680E+00 S2
 -7.1870470E-01 S1
 1.2812416E+01 S0

NUMERATOR R/ RUD
 -1.8124237E+00 S3
 -2.3369837E+00 S2
 1.1512384E+00 S1
 6.0457057E-01 S0

NUMERATOR P/ RUD
 6.1524420E+00 S3
 -4.8625565E-01 S2
 1.2697907E+01 S1
 -6.0138572E-02 S0

NUMERATOR BETA/ RUD
 1.9959368E-02 S3
 2.4346323E+00 S2
 2.4293251E+00 S1
 -5.4308750E-02 S0

NUMERATOR /

NUMERATOR AY/ RUD
 1.3465055E+01 S4
 6.9174805E+00 S3
 -9.6280853E+01 S2
 -8.9026611E+01 S1
 1.6901846E+00 S0

NUMERATOR AY/ AIL
 -1.0958076E+01 S4
 -1.6084473E+01 S3
 -1.1191016E+02 S2
 -1.0089063E+02 S1
 -4.9673920E+00 S0

AIRCRAFT STATE

MACH = .90000 ALT = 40000.00 VTRFPS = 871.65
 NZ = .99791 ALPHA = 4.58740 BETA = .00000
 GAMMA = .00000 THETA = 4.58742 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 Ixz = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -3.68992 FLAPS = 2.44223 STRAKES= -5.33240
 ALLERON= .00000 RUDDER = .00000
 THRUST = 1916.181 % RPM = 35.28230

DYNAMICS

CXAERO = -.01555936 CYAERO = .00000000 CZAERO = -.38450485
 CLMAERO= .00000000 CMMAERO= .00553221 CNMAERO= .00000000
 VC KTS = 277.196 DYN PR = 223.12401 RHO = .00058734

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.3232349	-6.0327034	1.7513971	.0000000
PITCH RATE(PER RAD)	.0000000	-8.0018721	-8.8413353	.0000000
STRAKE (PER RAD)	.0074185	-.1913485	-.2966666	.0000000
YACH (PER M #)	-.0791032	-.2048849	-.2091774	1189.0085449
ALPHA DOT (PER RAD)	.0000000	-.4485797	-1.2793598	.0000000
CANARD (PER RAD)	-.0152292	-.4517300	1.0257025	.0000000
FLAPERON (PER RAD)	.0309393	-1.5183449	-.6590127	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1040640	.2412131	-.0665323
ROLL RATE (PER RAD)	.0000000	-.0193545	-.3948224
YAW RATE (PER RAD)	.0000000	-.1612836	.2107756
AILERON (PER RAD)	-.2988185	.0539945	.1685113
RUDDER (PER RAD)	.2543252	-.1518290	-.0478503

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-506.3220215	10.4098425	26.1088867
ALPHADOT (PER R/S)	-1539734	-0317509	0000000
Q (PER R/S)	-2.7642975	-2208351	0000000
VEL (PER FT/S)	-0447719	-0022126	-0097530
STRAKE (PER RAD)	-15.9607983	-1.7891846	-6187908
FLAPRN (PER RAD)	-126.6485138	-3.9744797	2.5807581
CANARD (PER RAD)	-37.6797943	6.1859674	-1.2703056

A Matrix:

B Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-0.9753E-02	0.2611E+02	-0.6971E+02	-0.3207E+02	-0.6188E+00	-0.2581E+01	-0.1270E+01
-0.5152E-04	-0.5826E+00	0.9966E+00	-0.2961E-02	-0.1837E-01	-0.1457E+00	-0.4336E-01
-0.2211E-02	0.1043E+02	-0.2525E+00	0.9402E-04	-0.1789E+01	-0.3970E+01	-0.6187E+01
0.0000E+00	0.0000E+00	0.1000E+01	0.0000E+00	0.0000E+00	0.0000E+00	0.0000E+00

SYSTEM POLES

-3.6653728E+00	+JC	0.0000000E+00)
2.8410664E+00	+JC	0.0000000E+00)
-1.0276511E-02	+JC	7.4189186E-02)
-1.0276511E-02	+JC	-7.4189186E-02)

DYNAMICS

PERIOD= 84.6914	WD= 0.07419
ZETA= 0.13721	WN= 0.07490

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
8.4487182E-01 S3
-1.0390960E+01 S2
-2.0940661E-01 S1
-5.8414418E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
6.1879081E-01	S3	-1.8366557E-02	S3	-1.7886019E+00	S3	0.0000000E+00	S3
1.2472815E+02	S2	-1.7874441E+00	S2	-1.2524557E+00	S2	-1.7886019E+00	S2
9.0361526E+01	S1	-1.7097607E-02	S1	-1.4506564E-02	S1	-1.2524557E+00	S1
3.9721420E+01	S0	-1.5968226E-03	S0	4.6108522E-07	S0	-1.4506564E-02	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
2.5807581E+00	S3	-1.4573812E-01	S3	-3.9698524E+00	S3	0.0000000E+00	S3
2.7510498E+02	S2	-3.9948721E+00	S2	-3.8772240E+00	S2	-3.9698524E+00	S2
2.6381299E+02	S1	-2.4693001E-02	S1	-3.9018486E-02	S1	-3.8772240E+00	S1
1.2330817E+02	S0	3.9073564E-03	S0	1.2778273E-06	S0	-3.9018486E-02	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-1.2703056E+00	S3	-4.3359239E-02	S3	6.1873446E+00	S3	0.0000000E+00	S3
-4.3353882E+02	S2	6.1552610E+00	S2	3.2159796E+00	S2	6.1873446E+00	S2
-2.4449826E+02	S1	7.3440433E-02	S1	4.3892812E-02	S1	3.2159796E+00	S1
-1.0163135E+02	S0	1.3109546E-02	S0	-1.3361623E-06	S0	4.3892812E-02	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-1.5957973E+01	S4	-1.2662608E+02	S4	-3.7673126E+01	S4		
1.0058594E+00	S3	-2.1738770E+01	S3	-2.7875000E+01	S3		
1.0687512E+03	S2	3.3370959E+03	S2	-2.7145059E+03	S2		
7.9938745E+00	S1	2.7319473E+01	S1	-1.8470825E+01	S1		
-3.7729688E-02	S0	-1.0151470E-01	S0	1.1410826E-01	S0		

LATERAL-DIR. SYSTEM

30DY-AXIS STABILITY DERIVATIVES

	Y	L	N
	FT/SEC**2	RAD/SEC**2	RAD/SEC**2
BETA (PER RAD)	-92.0924225	-14.5608988	4.6504421
P (PER R/S)	.0000000	-1.5453510	-.0602025
R (PER R/S)	.0000000	.8014621	-.0259171
AILERON (PER RAD)	-24.9251251	42.6709595	2.6384134
RUDDER (PER RAD)	21.2139367	10.6608400	-2.8751526

A Matrix:

B Matrix:

BETA	P	R	PHI	AIL	RUD
-.1057E+00	.7998E-01	-.9968E+00	.3679E-01	-.2860E-01	.2434E-01
-.1456E+02	-.1545E+01	.8017E+00	.0000E+00	.4267E+02	.1066E+02
.4650E+01	-.6020E-01	-.2592E-01	.0000E+00	.2638E+01	-.2875E+01
.0000E+00	.1000E+01	.8024E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-1.1561203E-01	,J(2.3942432E+00)
-1.1561203E-01	,J(-2.3942432E+00)
-1.4631929E+00	,J(0.0000000E+00)
1.7495833E-02	,J(0.0000000E+00)

DYNAMICS

PERIOD= 2.6243	WD= 2.39424
ZETA= .04823	WN= 2.39703

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 1.6769199E+00 S3
 6.0544357E+00 S2
 8.3006887E+00 S1
 -1.4709139E-01 S0

NUMERATOR PHI/ AIL
 0.0000000E+00 S3
 4.2882645E+01 S2
 8.2783823E+00 S1
 2.3785645E+02 S0

NUMERATOR R/ AIL
 2.6384134E+00 S3
 1.6541500E+00 S2
 1.8872421E+01 S1
 8.7147017E+00 S0

NUMERATOR P/ AIL
 4.2670959E+01 S3
 8.1456594E+00 S2
 2.3634219E+02 S1
 -6.9941175E-01 S0

NUMERATOR BETA/ AIL
 -2.8595246E-02 S3
 7.3790431E-01 S2
 3.2933515E-01 S1
 1.2296468E-01 S0

NUMERATOR PHI/ RUD
 0.0000000E+00 S3
 1.0430148E+01 S2
 -1.6799297E+00 S1
 7.5774632E+00 S0

NUMERATOR R/ RUD
 -2.8751526E+00 S3
 -5.2755146E+00 S2
 2.7587050E-01 S1
 2.8377593E-01 S0

NUMERATOR P/ RUD
 1.0660840E+01 S3
 -1.2566423E+00 S2
 7.5553284E+00 S1
 -2.2769574E-02 S0

NUMERATOR BETA/ RUD
 2.4337485E-02 S3
 3.7568293E+00 S2
 5.2922964E+00 S1
 -8.9653969E-02 S0

NUMERATOR /

NUMERATOR AY/ RUD
 2.1145874E+01 S4
 2.2841797E+01 S3
 -2.3231607E+02 S2
 -3.1104053E+02 S1
 5.1327572E+00 S0

NUMERATOR AY/ AIL
 -2.4845276E+01 S4
 -4.1222656E+01 S3
 -2.1978247E+02 S2
 -2.3755859E+02 S1
 -7.6590424E+00 S0

AIRCRAFT STATE

MACH =	1.10000	ALT =	40000.00	VTREFS =	1065.35
NZ =	.99813	ALPHA =	3.86584	BETA =	.00000
GAMMA =	.00000	THETA =	3.86586	ACGW =	15926.00
IXX =	4548.00	IYY =	49429.00	IZZ =	52531.00
IXZ =	1327.00	XCG =	450.56006	ZCG =	64.88000
GEAR DOWN:	F	NY =	.00000	NX =	.00000
NORMAL MODE:	T	DR MODE:	F	AR MODE:	F

CONTROLS

CANARDS=	-2.89066	FLAPS =	-2.29115	STRAKES=	-7.68836
AILERON=	.00000	RUDDER =	.00000		
THRUST =	4104.059	% RPM =	62.32430		

DYNAMICS

CXAERO =	-.04913092	CYAERO =	.00000000	CZAERO =	-.25762522
CLMAERO=	.00000000	CMMAERO=	.00793071	CNMAERO=	.00000000
VC KTS =	350.281	DYN PR =	333.30859	RHO =	.00358734

A/C CONSTANTS

AREA =	185.0490	CBAR =	7.2200	SPAN =	27.2000
--------	----------	--------	--------	--------	---------

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.2597533	-6.1994114	.8500366	.0000000
PITCH RATE(PER RAD)	.0000000	-7.0105257	-9.1282816	.0000000
STRAKE (PER RAD)	.0104088	-.1765047	-.3105793	.0000000
MACH (PER M #)	-.0508395	.2583879	-.0228833	3000.6291504
ALPHA DOT (PER RAD)	.0000000	-.4071669	-1.5345602	.0000000
CANARD (PER RAD)	-.0288025	-.6750320	.9066220	.0000000
FLAPERON (PER RAD)	.0278860	-.8984005	-.6596896	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.1827812	.2385446	-.0894301
ROLL RATE (PER RAD)	.0000000	.0587365	-.4734246
YAW RATE (PER RAD)	.0000000	-.4056902	.2523658
AILERON (PER RAD)	-.3557255	.0734380	.1123293
RUDDER (PER RAD)	.1535030	-.0965280	.0310031

LONGITUDINAL SYSTEM

BODY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z FT/SEC**2	M RAD/SEC**2	X FT/SEC**2
ALPHA (PER RAD)	-770.8933105	7.6177168	30.9267120
ALPHADOT (PER R/S)	-1.1711344	-0.0466344	-0.0000000
Q (PER R/S)	-2.9600163	-0.2786695	-0.0000000
VEL (PER FT/S)	.0219260	-0.0006171	-0.0137817
STRAKE (PER RAD)	-21.9931030	-2.7980766	1.2969751
FLAPRN (PER RAD)	-111.9438171	-5.9432883	3.4746866
CANARD (PER RAD)	-84.1113129	8.1679564	-3.5888901

A Matrix:

B Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-1.1378E-01	.3093E+02	-7.183E+02	-.3210E+02	.1297E+01	.3475E+01	-.3589E+01
-.2062E-04	-.7251E+00	.9971E+00	-.2040E-02	-.2069E-01	-.1053E+00	-.7912E-01
-.6181E-03	.7652E+01	-.3252E+00	.9515E-04	-.2797E+01	-.5938E+01	.8172E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-3.3024406E+00 ,+J(0.0000000E+00)
2.2530622E+00 ,+J(0.0000000E+00)
-7.3574781E-03 ,+J(3.4283292E-02)
-7.3574781E-03 ,+J(-3.4283292E-02)

DYNAMICS

PERIOD=183.2725 W0= -.03428
ZETA= .20983 WN= .03506

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
1.0640841E+00 S3
-7.4238615E+00 S2
-1.0319161E-01 S1
-9.1476776E-03 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
1.2969751E+00	S3	-2.0687696E-02	S3	-2.7971115E+00	S3	0.0000000E+00	S3
2.0162851E+02	S2	-2.7953593E+00	S2	-2.2259293E+00	S2	-2.7971115E+00	S2
1.5079558E+02	S1	-2.8545175E-02	S1	-2.8330659E-02	S1	-2.2259293E+00	S1
7.0387589E+01	S0	2.3420961E-03	S0	5.8958735E-06	S0	-2.8330658E-02	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
3.4746866E+00	S3	-1.0529935E-01	S3	-5.9383733E+00	S3	0.0000000E+00	S3
4.2692432E+02	S2	-5.9565086E+00	S2	-5.1959218E+00	S2	-5.9383783E+00	S2
3.4792847E+02	S1	-5.8586456E-02	S1	-6.5655351E-02	S1	-5.1958218E+00	S1
1.6452283E+02	S0	6.1912909E-03	S0	1.2417004E-05	S0	-6.5655351E-02	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-3.5888901E+00	S3	-7.9118848E-02	S3	8.1716461E+00	S3	0.0000000E+00	S3
-5.9315527E+02	S2	8.1206884E+00	S2	5.4350100E+00	S2	8.1716461E+00	S2
-3.6672952E+02	S1	3.8854969E-02	S1	7.0660770E-02	S1	5.4350100E+00	S1
-1.7133297E+02	S0	-4.0726587E-03	S0	-1.7452505E-05	S0	7.0660770E-02	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-2.1989553E+01	S4	-1.1192581E+02	S4	-8.4097778E+01	S4		
1.3303105E+00	S3	-1.9273438E+01	S3	-5.4164063E+01	S3		
2.3295974E+03	S2	5.4476367E+03	S2	-5.6648633E+03	S2		
2.7774551E+01	S1	6.5097275E+01	S1	-6.7546896E+01	S1		
-6.7721069E-02	S0	-1.5561623E-01	S0	1.7182336E-01	S0		

LATERAL-DIR. SYSTEM
 BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-147.3787079	-30.3523254	6.5626097
P (PER R/S)	.0000000	-2.2511730	-.0543483
R (PER R/S)	.0000000	1.1378374	-.1258227
AILERON (PER RAD)	-44.3246613	42.9782257	3.8401041
RUDDER (PER RAD)	19.1270142	10.3424301	-2.7230482

B Matrix:

	P	R	PHI	AIL	RUD
BETA					
-1383E+00	.6742E-01	-.9977E+00	.3013E-01	-.4161E-01	.1795E-01
-3035E+02	-.2251E+01	.1138E+01	.0000E+00	.4298E+02	.1034E+02
.6563E+01	-.5435E-01	-.1258E+00	.0000E+00	.3840E+01	-.2723E+01
.0000E+00	.1000E+01	.6757E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-2.3190653E-01 +J(-2.8774700E+00)
 -2.3190653E-01 +J(-2.8774700E+00)
 -2.0598946E+00 +J(0.0000000E+00)
 8.3514936E-03 +J(0.0000000E+00)

DYNAMICS

PERIOD= 2.1836 WD= 2.87747
 ZETA= .03033 WN= 2.88680

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
 2.5153332E+00 S3
 9.2679510E+00 S2
 1.7088776E+01 S1
 -1.4336807E-01 S0

NUMERATOR PHI/ AIL
 0.0000000E+00 S3
 4.3237701E+01 S2
 1.7429138E+01 S1
 4.0072778E+02 S0

NUMERATOR R/ AIL
 3.8401041E+00 S3
 6.5671339E+00 S2
 2.7063492E+01 S1
 1.2010599E+01 S0

NUMERATOR PHI/ RUD
 0.0000000E+00 S3
 1.0158422E+01 S2
 -1.3809719E+00 S1
 -1.5036950E+01 S0

NUMERATOR R/ RUD
 -2.7230482E+00 S3
 -6.9510231E+00 S2
 -1.6272211E+00 S1
 -4.4527000E-01 S0

NUMERATOR /

NUMERATOR BETA/ AIL
 -4.1603581E-02 S3
 -1.0326662E+00 S2
 -4.3469572E+00 S1
 3.0744481E-01 S0

NUMERATOR BETA/ RUD
 1.7953675E-02 S3
 3.4568157E+00 S2
 6.8680439E+00 S1
 -6.7776084E-02 S0

NUMERATOR AY/ AIL
 -4.4223801E+01 S4
 -1.0285693E+02 S3
 -2.4806982E+02 S2
 -1.1754688E+02 S1
 -3.8948227E+01 S0

NUMERATOR AY/ RUD
 1.9083496E+01 S4
 3.7085205E+01 S3
 -3.4884424E+02 S2
 -6.8519213E+02 S1
 7.2460232E+00 S0

AIRCRAFT STATE

MACH = 1.30000 ALT = 40000.00 VTRFPS = 1259.05
 NZ = .99801 ALPHA = 3.49960 BETA = .00000
 GAMMA = .00000 THETA = 3.49962 ACGW = 15926.00
 IXX = 4548.00 IYY = 49429.00 IZZ = 52531.00
 Ixz = 1827.00 XCG = 450.56006 ZCG = 64.88000
 GEAR DOWN: F NY = .00000 NX = .00000
 NORMAL MODE: T DR MODE: F AR MODE: F

CONTROLS

CANARDS= -1.67758 FLAPS = -.84759 STRAKES= -8.86696
 AILERON= .00000 RUDDER = .00000
 THRUST = 6250.461 % RPM = 84.30560

DYNAMICS

CXAERO = -.06127162 CYAERO = .00000000 CZAERO = -.18452901
 CLMAERO= .00000000 CMMAERO= .00864378 CNMAERO= .00000000
 VC KTS = 423.966 DYN PR = 465.53027 PHO = .00058734

A/C CONSTANTS

AREA = 185.0490 CBAR = 7.2200 SPAN = 27.2000

LONGITUDINAL NONDIMENSIONAL DERIVATIVES

	CX BODY	CZ BODY	CM	THRUST(LBS)
ALPHA (PER RAD)	.1954049	-5.5153999	.3474554	.0000000
PITCH RATE (PER RAD)	.0000000	-5.6433468	-7.8051243	.0000000
STRAKE (PER RAD)	.0094373	-.1388975	-.2853293	.0000000
MACH (PER M #)	-.0281435	.1287043	-.2086425	4660.0000000
ALPHA DOT (PER RAD)	.0000000	-.3704265	-1.3850050	.0000000
CANARD (PER RAD)	-.0365331	-.5175204	.6851836	.0000000
FLAPERON (PER RAD)	.0200535	-.6250997	-.6357503	.0000000

LATERAL NONDIMENSIONAL DERIVATIVES

	CY BODY	CN	CL
BETA (PER RAD)	-1.2321653	.2525761	-.0917249
ROLL RATE (PER RAD)	.0000000	.0606781	-.4857363
YAW RATE (PER RAD)	.0000000	-.4446058	.2222201
AILERON (PER RAD)	-.2681391	.0969180	.0750780
RUDDER (PER RAD)	.0858089	-.0629783	.0191189

LONGITUDINAL SYSTEM
 90DY-AXIS DIMENSIONAL STABILITY DERIVATIVES

	Z	M	X
	FT/SEC**2	RAD/SEC**2	FT/SEC**2
ALPHA (PER RAD)	-958.4235840	4.1610947	32.1928406
ALPHADOT (PER R/S)	-1841514	-0.0497831	.0000000
Q (PER R/S)	-2.8159876	-2815988	.0000000
VEL (PER FT/S)	.0187018	-0.0028290	-.0138832
STRAKE (PER RAD)	-24.1727448	-3.5903320	1.6424055
FLAPRN (PER RAD)	-108.7878876	-7.9997215	3.4899712
CANARD (PER RAD)	-90.0655518	8.6217461	-6.3579636

A Matrix:

VEL	ALPHA	Q	THETA	STRAKE	FLAPRN	CANARD
-.1388E-01	.3219E+02	-.7685E+02	-.3211E+02	.1642E+01	.3490E+01	-.6358E+01
.1488E-04	-.7625E+00	.9976E+00	-.1563E-02	-.1923E-01	-.8655E-01	-.7166E-01
-.2830E-02	.4199E+01	-.3313E+00	.7779E-04	-.3589E+01	-.7995E+01	.8625E+01
.0000E+00	.0000E+00	.1000E+01	.0000E+00	.0000E+00	.0000E+00	.0000E+00

B Matrix:

SYSTEM POLES

-2.6458645E+00	+J(0.0000000E+00)
1.5918798E+00	+J(0.0000000E+00)
-2.6850533E-02	+J(1.2356019E-01)
-2.6850533E-02	+J(-1.2356019E-01)

DYNAMICS

PERIOD= 50.8512	WD= .12356
ZETA= .21235	WN= .12644

CHARACTERISTIC POLYNOMIAL

1.0000000E+00 S4
1.1076813E+00 S3
-4.1392813E+00 S2
-2.0933163E-01 S1
-6.7340255E-02 S0

NUMERATOR	U/STRK	NUMERATOR	ALP/STRK	NUMERATOR	Q/STRK	NUMERATOR	THT/STRK
1.6424055E+00	S3	-1.9232184E-02	S3	-3.5993745E+00	S3	0.0000000E+00	S3
2.7703589E+02	S2	-3.5874214E+00	S2	-2.8722610E+00	S2	-3.5893745E+00	S2
2.0988072E+02	S1	-4.0530451E-02	S1	-3.9088510E-02	S1	-2.8722610E+00	S1
9.0681473E+01	S0	3.5475290E-03	S0	2.9642897E-06	S0	-3.9088510E-02	S0
NUMERATOR	U/FLAP	NUMERATOR	ALP/FLAP	NUMERATOR	Q/FLAP	NUMERATOR	THT/FLAP
3.4899712E+00	S3	-8.6553216E-02	S3	-7.9954128E+00	S3	0.0000000E+00	S3
6.1551221E+02	S2	-8.0061522E+00	S2	-6.5811024E+00	S2	-7.9954128E+00	S2
4.8181912E+02	S1	-8.0498695E-02	S1	-8.5283577E-02	S1	-6.5811024E+00	S1
2.0788867E+02	S0	1.1873890E-02	S0	6.5767799E-06	S0	-8.5283577E-02	S0
NUMERATOR	U/CAN	NUMERATOR	ALP/CAN	NUMERATOR	Q/CAN	NUMERATOR	THT/CAN
-6.3579636E+00	S3	-7.1657479E-02	S3	8.6253138E+00	S3	0.0000000E+00	S3
-6.7215308E+02	S2	8.5799007E+00	S2	6.4139481E+00	S2	8.6253138E+00	S2
-4.5807202E+02	S1	1.2929159E-01	S1	1.0284895E-01	S1	6.4139481E+00	S1
-2.0202924E+02	S0	2.1762846E-03	S0	-7.1956465E-06	S0	1.0284895E-01	S0
NUMERATOR	AN/STRK	NUMERATOR	AN/FLAP	NUMERATOR	AN/CAN		
-2.4169205E+01	S4	-1.0877197E+02	S4	-9.0052399E+01	S4		
2.4531250E+00	S3	-1.3496094E+01	S3	-5.7070313E+01	S3		
3.5516045E+03	S2	8.1536445E+03	S2	-7.8810273E+03	S2		
4.7940033E+01	S1	1.0917351E+02	S1	-1.1391934E+02	S1		
-8.0492735E-02	S0	-1.7575693E-01	S0	2.1103203E-01	S0		

LATERAL-DIR. SYSTEM

BODY-AXIS STABILITY DERIVATIVES

	Y FT/SEC**2	L RAD/SEC**2	N RAD/SEC**2
BETA (PER RAD)	-214.4373322	-43.3369904	9.7589903
P (PER R/S)	.0000000	-2.7295818	-.0656977
R (PER R/S)	.0000000	1.1669350	-.1736327
AILERON (PER RAD)	-46.6650391	40.9900208	5.7486687
RUDDEP (PER RAD)	14.9335680	8.8452816	-2.5015326

A Matrix:

B Matrix:

BETA	P	R	PHI	AIL	RUD
-.1703E+00	.6104E-01	-.9981E+00	.2551E-01	-.3706E-01	.1186E-01
-.4334E+02	-.2730E+01	.1167E+01	.0000E+00	.4099E+02	.8845E+01
.9759E+01	-.6570E-01	-.1736E+00	.0000E+00	.5749E+01	-.2502E+01
.0000E+00	.1000E+01	.6116E-01	.0000E+00	.0000E+00	.0000E+00

SYSTEM POLES

-2.7606922E-01 +J(-3.4604406E+00)
-2.7606922E-01 +J(-3.4604406E+00)
-2.5261230E+00 +J(0.0000000E+00)
4.7477148E-03 +J(0.0000000E+00)

DYNAMICS

PERIOD= 1.8157 WD= 3.46044
ZETA= .07953 WN= 3.47144

CHARACTERISTIC POLYNOMIAL

1.000000E+00 S4
 3.073530E+00 S3
 1.343120E+01 S2
 3.037820E+01 S1
 -1.445521E-01 S0

NUMERATOR BETA/ AIL	NUMERATOR P/ AIL	NUMERATOR R/ AIL	NUMERATOR PHI/ AIL
-3.706356E-02 S3	4.0990021E+01 S3	5.7486687E+00 S3	0.000000E+00 S3
-3.343470E+00 S2	2.241299E+01 S2	1.361589E+01 S2	4.134156E+01 S2
-1.109627E+01 S1	6.501521E+02 S1	4.074601E+01 S1	2.324566E+01 S1
3.729204E-01 S0	-1.013443E+00 S0	1.655744E+01 S0	6.526440E+02 S0

NUMERATOR BETA/ RUD	NUMERATOR P/ RUD	NUMERATOR R/ RUD	NUMERATOR PHI/ RUD
1.186094E-02 S3	8.845281E+00 S3	-2.501532E+00 S3	0.000000E+00 S3
3.071229E+00 S2	-3.908233E-01 S2	-7.719553E+00 S2	8.692298E+00 S2
7.539236E+00 S1	-2.223646E+01 S1	-2.260463E+00 S1	-8.629147E-01 S1
-4.684279E-02 S0	3.435192E-02 S0	-5.633542E-01 S0	-2.237469E+01 S0

NUMERATOR AY/ AIL /

NUMERATOR AY/ RUD
 1.490572E+01 S4
 3.614379E+01 S3
 -4.757136E+02 S2
 -1.162928E+03 S1
 7.929977E+00 S0